

alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To correct cracking in the axle adapter of the shock absorber of the NLG, which could cause failure of the NLG and consequent damage to the airplane structure, accomplish the following:

(a) Within 300 flight hours after the effective date of this AD, perform a one-time visual inspection to detect cracking in the axle adapter of the NLG shock absorber, in accordance with Dornier Service Bulletin SB-328-32-213, dated April 16, 1997.

(1) If no cracking is detected, no further action is required by this AD.

(2) If any cracking is detected, prior to further flight, remove the NLG shock absorber and replace it with a new or serviceable part, in accordance with the service bulletin.

Note 2: Dornier Service Bulletin SB-328-32-213, dated April 16, 1997, references Messier-Dowty Service Bulletin 800-32-027, dated May 7, 1997, as an additional source of service information to accomplish the inspection, removal, and repair.

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their request through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(d) The actions shall be done in accordance with Dornier Service Bulletin SB-328-32-213, dated April 16, 1997. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from FAIRCHILD DORNIER, DORNIER Luftfahrt GmbH, P.O. Box 1103, D-82230 Wessling, Germany. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC 20401.

Note 4: The subject of this AD is addressed in German airworthiness directive 97-142, dated May 22, 1997.

(e) This amendment becomes effective on August 12, 1998.

Issued in Renton, Washington, on June 29, 1998.

Vi L. Lipski,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 98-17913 Filed 7-7-98; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 98-NM-132-AD; Amendment 39-10646; AD 98-14-13]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A300, A310, A300-600 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Airbus Model A300, A310, and A300-600 series airplanes. This amendment requires a one-time operational test and repetitive functional tests of the free fall control mechanism of the landing gear to ensure proper release of the main landing gear (MLG), and corrective action, if necessary. This amendment also requires eventual modification of the free fall control mechanism of the landing gear, which constitutes terminating action for the repetitive functional tests. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to prevent malfunction of the free fall control mechanism of the landing gear, which could result in the inability to extend the MLG in the event of failure of the hydraulic extension system.

DATES: Effective August 12, 1998.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of August 12, 1998.

ADDRESSES: The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Airbus Model A300, A310, and A300-600 series airplanes was published in the **Federal Register** on May 14, 1998 (63 FR 26742). That action proposed to require a one-time operational test and repetitive functional tests of the free fall control mechanism of the landing gear to ensure proper release of the main landing gear (MLG), and corrective action, if necessary. That action also proposed to require eventual modification of the free fall control mechanism of the landing gear, which constitutes terminating action for the repetitive functional tests.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

The FAA estimates that 24 Model A300 series airplanes, 41 Model A310 series airplanes, and 61 Model A300-600 series airplanes of U.S. registry will be affected by this AD.

It will take approximately 3 work hours per airplane to accomplish the required operational test, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the required operational test on U.S. operators is estimated to be \$22,680, or \$180 per airplane.

It will take approximately 2 work hours per airplane to accomplish the required functional test, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the required functional test on U.S. operators is estimated to be \$15,120, or \$120 per airplane, per test cycle.

It will take approximately 26 work hours per airplane to accomplish the required modification on the Model A300 and A300-600 series airplanes, at an average labor rate of \$60 per work hour. Required parts will cost

approximately \$2,630 per airplane. Based on these figures, the cost impact of the required modification on U.S. operators of Model A300 or A300-600 series airplanes is estimated to be \$356,150, or \$4,190 per airplane.

It will take approximately 28 work hours per airplane to accomplish the required modification on the Model A310 series airplanes, at an average labor rate of \$60 per work hour. Required parts will cost approximately \$3,710 per airplane. Based on these figures, the cost impact of the required modification on U.S. operators of Model A310 series airplanes is estimated to be \$220,990, or \$5,390 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

98-14-13 Airbus Industrie: Amendment 39-10646. Docket 98-NM-132-AD.

Applicability: Model A300, A310, and A300-600 series airplanes; on which Airbus Industrie Modification 02781 has been accomplished, and on which Airbus Industrie Modification 03433 or 04443 has not been accomplished; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent malfunction of the free fall control mechanism of the landing gear, which could result in the inability to extend the main landing gear (MLG) in the event of failure of the hydraulic extension system, accomplish the following:

(a) Within 600 flight hours after the effective date of this AD, perform a one-time operational test of the free fall control mechanism of the landing gear to ensure proper release of the MLG for extension by free fall, in accordance with Airbus Industrie All Operator Telex (AOT) 32-14, dated February 3, 1997, or Revision 01, dated March 13, 1997. If any discrepancy is detected in the functioning of the free fall control mechanism of the landing gear, prior to further flight, readjust the mechanism, and repeat the operational test in accordance with the AOT. If any discrepancy is detected in the second operational test, prior to further flight, rereg the free fall control mechanism in accordance with the AOT, and accomplish the actions required by paragraph (b) of this AD.

(b) Within 10 months after the effective date of this AD, perform a functional test of the free fall control mechanism of the landing gear to ensure proper release of the MLG for extension by free fall, in accordance with Airbus Industrie AOT 32-14, dated February 3, 1997, or Revision 01, dated March 13, 1997. Thereafter, repeat the functional test of the free fall control mechanism of the landing gear at intervals not to exceed 12 months,

until the modification required by paragraph (c) of this AD has been accomplished. During any test performed in accordance with paragraph (b) of this AD, if the free fall control mechanism of the landing gear fails to fully extend the MLG, prior to further flight, readjust or rereg the mechanism in accordance with the AOT.

(c) Within 66 months after the effective date of this AD, modify the free fall control mechanism of the landing gear in accordance with Airbus Industrie Service Bulletin A300-32-0425, Revision 01 (for Model A300 series airplanes); A310-32-2111, Revision 01 (for Model A310 series airplanes); or A300-32-6072, Revision 01 (for Model A300-600 series airplanes); all dated October 10, 1997; as applicable. Accomplishment of the modification constitutes terminating action for the repetitive functional tests required by paragraph (b) of this AD.

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(f) The actions shall be done in accordance with the following Airbus Industrie service information, as applicable:

- All Operator Telex (AOT) 32-14, dated February 3, 1997;
- All Operator Telex (AOT) 32-14, Revision 01, dated March 13, 1997;
- Service Bulletin A300-32-0425, Revision 01, dated October 10, 1997;
- Service Bulletin A310-32-2111, Revision 01, dated October 10, 1997; or
- Service Bulletin A300-32-6072, Revision 01, dated October 10, 1997.

This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 3: The subject of this AD is addressed in French airworthiness directive 97-113-221(B)R1, dated December 3, 1997.

(g) This amendment becomes effective on August 12, 1998.

Issued in Renton, Washington, on June 29, 1998.

Vi L. Lipski,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 98-17912 Filed 7-7-98; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 98-NM-93-AD; Amendment 39-10644; AD 98-14-11]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A319, A320, and A321 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to all Airbus Model A319, A320, and A321 series airplanes, that requires repetitive inspections for discrepancies of the lock bolt for the pintle pin on the main landing gear (MLG), and follow-on corrective actions, if necessary. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to detect and correct a rotated, damaged, or missing lock bolt, which could result in disengagement of the pintle pin from the bearing, and consequent collapse of the MLG during landing.

DATES: Effective August 12, 1998.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of August 12, 1998.

ADDRESSES: The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all Airbus Model A319, A320, and A321 series airplanes was published in the **Federal Register** on May 12, 1998 (63 FR 26111). That action proposed to require repetitive inspections for discrepancies of the lock bolt for the pintle pin on the main landing gear (MLG), and follow-on corrective actions, if necessary.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received.

The commenters support the proposed rule.

Explanation of Changes Made to This Final Rule

In the proposal, the FAA inadvertently omitted reference to Revision 1, dated June 13, 1994, of Airbus Service Bulletin A320-32-1119. Therefore, the FAA has revised the final rule accordingly.

Conclusion

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule with the change previously described. The FAA has determined that this change will neither increase the economic burden on any operator or increase the scope of the AD.

Interim Action

This is considered to be interim action until final action is identified, at which time the FAA may consider further rulemaking.

Cost Impact

The FAA estimates that 120 airplanes of U.S. registry will be affected by this AD. It will take approximately 1 work hour per airplane to accomplish the required inspection, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the inspection required by this AD on U.S. operators is estimated to be \$7,200, or \$60 per airplane, per inspection cycle.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

98-14-11 Airbus Industrie: Amendment 39-10644. Docket 98-NM-93-AD.

Applicability: All Model A319, A320, and A321 series airplanes; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in