

Service Bulletin F27/30-44, dated February 20, 1998. Prior to further flight thereafter, revise the FAA-approved AFM to incorporate the flight manual changes described in Fokker MCNO F27-004, dated February 10, 1998.

(2) Install a modified pressure switch in the monitoring system in accordance with Fokker Service Bulletin F27/30-45, dated August 11, 1999.

Alternative Methods of Compliance

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be

used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

Special Flight Permits

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(f) Except as provided by paragraph (a) of this AD, the actions shall be done in accordance with the following Fokker service bulletins, including Fokker manual change notifications, as applicable, which contain the following specified effective pages:

Document referenced and date	Page No.	Revision level shown on page	Date shown on page
SBF50-30-025, Revision 2, Oct. 21, 1998	1-3, 14, 15, 17-19 4-13, 16, 22-25, 27-41, 43-45. 20, 21, 26, 42	2 Original 1	Oct. 21, 1998. Oct. 31, 1997. June 10, 1998.
Manual Change Notification MCNO F50-001			
SBF27/30-44, February 20, 1998	1 1-31	Original Original	Oct. 23, 1997. Feb. 20, 1998.
Manual Change Notification MCNM F27-003			
	1-5	Original	Dec. 10, 1997.
Manual Change Notification MCNO F27-004			
SBF50-30-026, Aug. 11, 1999	1-2	Original	Feb. 10, 1998.
F27/30-45, August 11, 1999	1-10	Original	Aug. 11, 1999.
	1-12	Original	Aug. 11, 1999.

This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Fokker Services B.V., P.O. Box 231, 2150 AE Nieuw-Vennep, the Netherlands. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 3: The subject of this AD is addressed in Dutch airworthiness directives 1998-019/2, and 1997-113/3, both dated June 18, 1999.

(g) This amendment becomes effective on April 12, 2000.

Issued in Renton, Washington, on February 29, 2000.

Donald L. Rigglin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.
[FR Doc. 00-5334 Filed 3-7-00; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-NM-241-AD; Amendment 39-11613; AD 2000-05-04]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A330 and A340 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Airbus Model A330 and A340 series airplanes, that requires repetitive inspections to detect cracking of the fuselage skin in the area of the VHF2 antenna, and repair, if necessary. This amendment also provides for optional terminating action for the repetitive inspections. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil

airworthiness authority. The actions specified by this AD are intended to detect and correct such cracking, which could result in cabin depressurization of the airplane.

DATES: Effective April 12, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of April 12, 2000.

ADDRESSES: The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601

Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Airbus Model A330 and A340 series airplanes was published in the **Federal Register** on December 30, 1999 (64 FR 73441). That action proposed to require repetitive inspections to detect cracking of the fuselage skin in the area of the VHF2 antenna, and repair, if necessary. That action also proposed to provide for optional terminating action for the repetitive inspections.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

None of the airplanes affected by this action are on the U.S. Register. All airplanes included in the applicability of this rule currently are operated by non-U.S. operators under foreign registry; therefore, they are not directly affected by this AD action. However, the FAA considers that this rule is necessary to ensure that the unsafe condition is addressed in the event that any of these subject airplanes are imported and placed on the U.S. Register in the future.

Should an affected airplane be imported and placed on the U.S. Register in the future, it will require approximately 6 work hours to accomplish the required inspections, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of this AD will be \$360 per airplane, per inspection cycle.

Should an operator elect to accomplish the optional terminating action rather than continue the repetitive inspections, it will take approximately 112 work hours per airplane to accomplish the modification, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of this optional terminating action is estimated to be \$6,720 per airplane.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects

14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

2000-05-04 Airbus: Amendment 39-11613. Docket 99-NM-241-AD.

Applicability: Model A330 and A340 series airplanes, certificated in any category; except those on which Airbus production modification 46025 is installed or on which Airbus Service Bulletin A330-53-3097, Revision 01, dated May 21, 1999 (for Model A330 series airplanes), or Service Bulletin A340-53-4108, Revision 01, dated May 21, 1999 (for Model A340 series airplanes), has been accomplished.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been

otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (g) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect and correct cracking of the fuselage skin in the area of the VHF2 antenna, which could result in cabin depressurization of the airplane, accomplish the following:

Detailed Visual Inspection

(a) At the latest of the times specified in paragraphs (a)(1), (a)(2), (a)(3), and (a)(4) of this AD, as applicable: Perform a detailed visual inspection (without removal of the VHF2 antenna) of the fuselage skin aft of frame 54, between the airplane centerline and stringer 56R in the area of the VHF2 antenna to detect cracks, in accordance with Airbus Service Bulletin A330-53-3094, Revision 02, dated May 28, 1998 (for Model A330 series airplanes), or Service Bulletin A340-53-4105, Revision 02, dated May 25, 1998 (for Model A340 series airplanes) (hereinafter referred to as the applicable service bulletin). Thereafter, if no cracks are detected, repeat the detailed visual inspection every 36 flight hours until accomplishment of the high frequency eddy current (HFEC) inspection required by paragraph (b) of this AD.

(1) Prior to the accumulation of 900 total flight hours.

(2) Within 1,250 flight hours since accomplishment of the interim repair specified by paragraph 2.C.(4) of the applicable service bulletin, if the interim repair has been accomplished prior to the effective date of this AD.

(3) Within 300 flight hours since the most recent HFEC inspection accomplished in accordance with the applicable service bulletin, if the most recent HFEC inspection has been accomplished prior to the effective date of this AD.

(4) Within 36 flight hours after the effective date of this AD.

Note 2: For the purposes of this AD, a detailed visual inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

High Frequency Eddy Current Inspection

(b) Perform a high frequency eddy current (HFEC) inspection to detect cracks of the fuselage skin aft of frame 54, between the

airplane centerline and stringer 56R in the area of the VHF2 antenna, in accordance with the applicable service bulletin, at the applicable time specified by paragraph (b)(1) or (b)(2) of this AD. Accomplishment of this inspection terminates the requirements of paragraph (a) of this AD.

(1) For airplanes on which the interim repair specified by paragraph 2.C.(4) of the applicable service bulletin has not been accomplished: Prior to the accumulation of 900 total flight hours on the airplane, or within 500 flight hours after the effective date of this AD, whichever occurs later. Thereafter, accomplish the follow-on actions of paragraph (c) or (d) of this AD, as applicable.

(2) For airplanes on which the interim repair specified by paragraph 2.C.(4) of the applicable service bulletin has been accomplished: Within 1,250 flight hours after accomplishment of the interim repair, or within 500 flight hours after the effective date of this AD, whichever occurs later.

Repetitive Inspections

(c) If no crack is detected during the HFEC inspection required by paragraph (b) of this AD, accomplish the repetitive inspections required by paragraph (c)(1) or (c)(2) of this AD, as applicable.

(1) For airplanes on which the interim repair specified by paragraph 2.C.(4) of the applicable service bulletin has not been accomplished, accomplish the actions specified by paragraphs (c)(1)(i) and (c)(1)(ii) of this AD.

(i) Repeat the HFEC inspection specified by paragraph (b) at intervals not to exceed 500 flight hours.

(ii) Within 300 flight hours after each HFEC inspection required by this AD: Perform a detailed visual inspection (without removal of the VHF2 antenna) of the fuselage skin aft of frame 54, between the airplane centerline and stringer 56R in the area of the VHF2 antenna to detect cracks, in accordance with the applicable service bulletin. Thereafter, if no cracks are detected, repeat the detailed visual inspection every 36 flight hours until accomplishment of the next HFEC inspection required by paragraph (c)(1)(i) of this AD.

(2) For airplanes on which the interim repair specified by paragraph 2.C.(4) of the applicable service bulletin has been accomplished, repeat the HFEC inspection specified by paragraph (b) of this AD at intervals not to exceed 1,250 flight hours.

Corrective Actions

(d) If any crack is detected during any inspection required by paragraph (a), (b), or (c) of this AD, and the interim repair specified by paragraph 2.C.(4) of the applicable service bulletin has not been accomplished: Prior to further flight, accomplish the actions specified by paragraph (d)(1) or (d)(2) of this AD, as applicable.

(1) If only one crack is detected and that crack is 9.45 inches or less, and is within the limits specified by the applicable service bulletin: Install the interim repair specified in paragraph 2.C.(4) of the applicable service bulletin. Thereafter, repeat the HFEC

inspection specified by paragraph (b) of this AD at intervals not to exceed 1,250 flight hours.

Note 3: The interim repair referenced by this AD consists of cutting out the cracked portion of the fuselage skin, and installing a filler plate in the skin cutout, two doublers, and shims, as described in paragraph 2.C.(4) of the applicable service bulletin.

Note 4: Accomplishment of the interim repair in accordance with paragraph 4.3 of Airbus Industrie All Operator Telex (AOT) 53-10, dated September 24, 1997, is acceptable for compliance with the requirements of paragraph (d)(1) of this AD.

(2) If any crack is detected that is longer than 9.45 inches, or is outside the limits specified by the service bulletin, or if more than one crack is detected: Repair in accordance with a method approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate; or the Direction Generale de l'Aviation Civile (DGAC) (or its delegated agent). For a repair method to be approved by the Manager, International Branch, ANM-116, as required by this paragraph, the Manager's approval letter must specifically reference this AD.

(e) If any crack is detected during any inspection required by paragraph (a), (b), or (c) of this AD and the interim repair specified by paragraph 2.C.(4) of the applicable service bulletin has been accomplished: Prior to further flight, repair in accordance with a method approved by the Manager, International Branch, ANM-116; or the DGAC (or its delegated agent). For a repair method to be approved by the Manager, International Branch, ANM-116, as required by this paragraph, the Manager's approval letter must specifically reference this AD.

(f) Accomplishment of the modification as described in Airbus Service Bulletin A330-53-3097, Revision 01, dated May 21, 1999 (for Model A330 series airplanes), or Service Bulletin A340-53-4108, Revision 01, dated May 21, 1999 (for Model A340 series airplanes), terminates the repetitive inspections required by paragraphs (a), (b), and (c) of this AD.

Note 5: Accomplishment of Airbus production modification 46025, or the modification as described in Airbus Service Bulletin A330-53-3097, dated July 29, 1998 (for Model A330 series airplanes), or Service Bulletin A340-53-4108, dated July 31, 1998 (for Model A340 series airplanes), also constitutes terminating action for the repetitive inspections required by paragraphs (a), (b), and (c) of this AD.

Alternative Methods of Compliance

(g) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 6: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be

obtained from the International Branch, ANM-116.

Special Flight Permits

(h) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(i) Except as provided by paragraphs (d)(2) and (e) of this AD, the actions shall be done in accordance with Airbus Service Bulletin A330-53-3094, Revision 02, dated May 28, 1998; or Airbus Service Bulletin A340-53-4105, Revision 02, dated May 25, 1998; as applicable. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 7: The subject of this AD is addressed in French airworthiness directives 1998-192-071(B)R1 (for Model A330 series airplanes) and 1998-193-089(B)R1 (for Model A340 series airplanes), both dated March 24, 1999.

(j) This amendment becomes effective on April 12, 2000.

Issued in Renton, Washington, on February 29, 2000.

Donald L. Riggins,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-NM-337-AD; Amendment 39-11616; AD 2000-05-07]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A300 and A300-600 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment supersedes an existing airworthiness directive (AD), applicable to certain Airbus Model A300 and A300-600 series airplanes, that currently requires repetitive inspections to detect cracks in Gear Rib 5 of the main landing gear (MLG) attachment fittings at the lower flange, and repair, if necessary. That AD also