

Special Flight Permits

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(f) The actions shall be done in accordance with Airbus All Operator Telex (AOT) 32-17, Revision 01, dated November 6, 1997; Airbus Service Bulletin A320-32-1187, dated June 17, 1998; or Airbus Service Bulletin A320-32-1187, Revision 01, dated February 17, 1999.

(1) The incorporation by reference of Airbus Service Bulletin A320-32-1187, dated June 17, 1998, and Airbus Service Bulletin A320-32-1187, Revision 01, dated February 17, 1999, is approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The incorporation by reference of Airbus All Operator Telex (AOT) 32-17, Revision 01, dated November 6, 1997, was approved previously by the Director of the Federal Register as of August 12, 1998 (63 FR 36834, July 8, 1998).

(3) Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 4: The subject of this AD is addressed in French airworthiness directive 97-385-112(B)R1, dated October 21, 1998.

(g) This amendment becomes effective on June 30, 2000.

Issued in Renton, Washington, on May 16, 2000.

Donald L. Riggins,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 00-12815 Filed 5-25-00; 8:45 am]

BILLING CODE 4910-13-U

DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. 97-NM-88-AD; Amendment 39-11748; AD 2000-10-23]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 747-100, -200, -300, 747SR, and 747SP Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment supersedes an existing airworthiness directive (AD), applicable to certain Boeing Model 747-100, -200, -300, 747SR, and 747SP

series airplanes, that currently requires a one-time inspection to detect cracking of the longeron splice fittings at stringer 11, on the left and right sides at body station 2598, and replacement of any cracked fitting with a new fitting. This amendment reduces the compliance time for accomplishment of the currently required inspection and adds a new requirement for repetitive inspections. This amendment is prompted by reports that fatigue cracking was found on longeron splice fittings. The actions specified by this AD are intended to detect and correct such fatigue cracking, which could result in reduced controllability of the horizontal stabilizer.

DATES: Effective June 30, 2000.

The incorporation by reference of Boeing Alert Service Bulletin 747-53A2410, Revision 2, including Addendum, dated October 30, 1997, as listed in the regulations, is approved by the Director of the Federal Register as of June 30, 2000.

The incorporation by reference of Boeing Service Bulletin 747-53A2410, Revision 3, including Addendum, dated March 12, 1998, as listed in the regulations, was approved previously by the Director of the Federal Register as of January 13, 1998 (62 FR 67550, December 29 1997).

ADDRESSES: The service information referenced in this AD may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC. **FOR FURTHER INFORMATION CONTACT:** Rick Kawaguchi, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Transport Airplane Directorate, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-1153; fax (425) 227-1181.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) by superseding AD 97-26-21, amendment 39-10264 (62 FR 67550, December 29 1997), which is applicable to certain Boeing Model 747-100, -200, -300, 747SR, and 747SP series airplanes, was published in the **Federal Register** on February 22, 2000 (65 FR 8667). The action proposed to continue to require a one-time inspection to detect cracking of the longeron splice fittings at stringer 11, on the left and

right sides at body station 2598, and replacement of any cracked fitting with a new fitting. The action also proposed to reduce the compliance time for accomplishment of the currently required inspection and add a new requirement for repetitive inspections.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

There are approximately 685 airplanes of the affected design in the worldwide fleet. The FAA estimates that 99 airplanes of U.S. registry will be affected by this AD.

The inspection that is currently required by AD 97-26-21 takes approximately 32 work hours per airplane to accomplish, at an average labor rate of \$60 per work hour.

Based on these figures, the cost impact of the currently required actions on U.S. operators is estimated to be \$190,080, or \$1,920 per airplane.

This AD requires the same inspection currently required by AD 97-26-21 to be accomplished repetitively. Therefore, the cost impact of the requirements of this AD on U.S. operators is estimated to be \$190,080, or \$1,920 per airplane, per inspection cycle.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT

Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by removing amendment 39–10264 (62 FR 67550, December 29 1997), and by adding a new airworthiness directive (AD), amendment 39–11748, to read as follows:

2000–10–23 Boeing: Amendment 39–11748.

Docket 97–NM–88–AD. Supersedes AD 97–26–21, Amendment 39–10264.

Applicability: Model 747–100, 747–300, 747SR, and 747SP series airplanes; having line positions 201 through 886 inclusive; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect and correct fatigue cracking of the longeron splice fittings at stringer 11, which could result in reduced controllability of the horizontal stabilizer, accomplish the following:

Initial Inspection

(a) Perform a one-time detailed visual inspection to detect cracking of the longeron fittings at stringer 11, on the left and right sides at body station 2598, at the time specified in paragraph (a)(1) or (a)(2) of this AD, as applicable, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747–53A2410, Revision 2, dated October 30, 1997, including Addendum; or Boeing Service Bulletin 747–53A2410, Revision 3, dated March 12, 1998, including Addendum. After the effective date of this AD, only Revision 3 shall be used.

(1) For airplanes that have accumulated fewer than 17,000 total flight cycles or 63,000 total flight hours as of the effective date of this AD: Inspect at the later of the times specified in paragraph (a)(1)(i) or (a)(1)(ii) of this AD.

(i) Prior to the accumulation of 17,000 total flight cycles or 63,000 total flight hours, whichever occurs first.

(ii) Within 1,800 flight cycles or 7,000 flight hours after the effective date of this AD, whichever occurs first.

(2) For airplanes that have accumulated 17,000 total flight cycles or more, or 63,000 total flight hours or more, as of the effective date of this AD: Inspect at the earlier of the times specified in paragraphs (a)(2)(i) and (a)(2)(ii) of this AD.

(i) Prior to the accumulation of 22,000 total flight cycles or 78,000 total flight hours, whichever occurs first.

(ii) Within 1,800 flight cycles or 7,000 flight hours after the effective date of this AD, whichever occurs first.

Note 2: Where there are differences between the AD and the service bulletin, the AD prevails.

Note 3: For the purposes of this AD, a detailed visual inspection is defined as: “An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required.”

Repetitive Inspections

(b) If no crack is found during the inspection required by paragraph (a) of this AD, repeat the inspection one time at the later of the times specified in paragraphs (b)(1) and (b)(2) of this AD, and thereafter at intervals not to exceed 3,000 flight cycles or 18,000 flight hours, whichever occurs first.

(1) Within 3,000 flight cycles or 18,000 flight hours after accomplishment of the most recent inspection, whichever occurs first.

(2) Within 1,800 flight cycles or 7,000 flight hours after the effective date of this AD, whichever occurs first.

Replacement and Repetitive Inspections

(c) If any crack is found during any inspection required by paragraph (a) or (b) of this AD: Prior to further flight, replace the cracked fitting with a new fitting, in accordance with the Accomplishment

Instructions of Boeing Alert Service Bulletin 747–53A2410, Revision 2, dated October 30, 1997, including Addendum; or Boeing Service Bulletin 747–53A2410, Revision 3, dated March 12, 1998, including Addendum. After the effective date of this AD, only Revision 3 shall be used. Then, repeat the inspection specified in paragraph (a) of this AD at the later of the times specified in paragraphs (c)(1) and (c)(2) of this AD, and thereafter at intervals not to exceed 3,000 flight cycles or 18,000 flight hours, whichever occurs first.

(1) Within 17,000 flight cycles or 63,000 flight hours after replacement, whichever occurs first.

(2) Within 1,800 flight cycles or 7,000 flight hours after the effective date of this AD, whichever occurs first.

Alternative Methods of Compliance

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

Note 4: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permits

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(f) The actions shall be done in accordance with Boeing Alert Service Bulletin 747–53A2410, Revision 2, including Addendum, dated October 30, 1997; or Boeing Service Bulletin 747–53A2410, Revision 3, including Addendum, dated March 12, 1998.

(1) The incorporation by reference of Boeing Service Bulletin 747–53A2410, Revision 3, including Addendum, dated March 12, 1998, is approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The incorporation by reference of Boeing Alert Service Bulletin 747–53A2410, Revision 2, including Addendum, dated October 30, 1997, was approved previously by the Director of the Federal Register as of January 13, 1998 (62 FR 67550, December 29 1997).

(3) Copies may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124–2207. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(g) This amendment becomes effective on June 30, 2000.

Issued in Renton, Washington, on May 18, 2000.

Donald L. Riggins,

*Acting Manager, Transport Airplane
Directorate, Aircraft Certification Service.*

[FR Doc. 00-13086 Filed 5-25-00; 8:45 am]

BILLING CODE 4910-13-U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2000-NM-111-AD; Amendment
39-11745; AD 2000-10-21]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 737 Series Airplanes

AGENCY: Federal Aviation
Administration, DOT.

ACTION: Final rule; request for
comments.

SUMMARY: This amendment adopts a new airworthiness directive (AD) that is applicable to certain Boeing Model 737 series airplanes. This action requires a one-time general visual inspection of the seat locks and seat tracks of the flightcrew seats to ensure that the seats lock in position and to verify that lock nuts and bolts of adequate length are installed on the rear tracklock bracket, and corrective action, if necessary. This action is necessary to prevent uncommanded movement of the flightcrew seats during acceleration and take-off of the airplane, which could result in reduced controllability of the airplane. This action is intended to address the identified unsafe condition.

DATES: Effective June 12, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of June 12, 2000.

Comments for inclusion in the Rules Docket must be received on or before July 25, 2000.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2000-NM-111-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: 9-anm-iarcomment@faa.gov. Comments sent

via fax or the Internet must contain "Docket No. 2000-NM-111-AD" in the subject line and need not be submitted in triplicate.

The service information referenced in this AD may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Keith Ladderud, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington; telephone (425) 227-2780; fax (425) 227-1181.

SUPPLEMENTARY INFORMATION: The FAA has received reports indicating instances of the pilot seat sliding to the aft-most position during acceleration and take-off on certain Boeing Model 737 series airplanes. Investigation revealed that the screws attaching the rear tracklock bracket to the seat track had broken, allowing excessive lateral movement and disengagement of the locking pin from the floor-mounted seat track. A disengaged locking pin can cause misalignment of the seat tracks. Such misalignment of the seat tracks, if not corrected, could result in uncommanded movement of the flightcrew seats during acceleration and take-off of the airplane, which could result in reduced controllability of the airplane.

Explanation of Relevant Service Information

The FAA has reviewed and approved Boeing Alert Service Bulletin 737-25A1363, dated November 5, 1998, which describes procedures for a one-time general visual inspection of the seat locks and seat tracks of the flightcrew seats to ensure that the seats lock in position and to verify that lock nuts and bolts of adequate length are installed on the rear tracklock bracket. If lock nuts and bolts of adequate length are not installed on the rear tracklock bracket, the service bulletin describes installation of lock nuts and bolts of adequate length on the tracklock bracket, and re-alignment of the seat tracks. Accomplishment of the actions specified in the alert service bulletin is intended to adequately address the identified unsafe condition.

Explanation of the Requirements of the Rule

Since an unsafe condition has been identified that is likely to exist or

develop on other Boeing Model 737 series airplanes of the same type design, this AD is being issued to prevent uncommanded movement of the flightcrew seats during acceleration and take-off of the airplane, and consequent reduced controllability of the airplane. This AD requires a one-time general visual inspection of the seat locks and seat tracks of the flightcrew seats to ensure that the seats lock in position and to verify that lock nuts and bolts of adequate length are installed on the rear tracklock bracket, and corrective action, if necessary. The actions are required to be accomplished in accordance with the alert service bulletin described previously, except as discussed below.

Difference Between Alert Service Bulletin and This AD

Operators should note that, although the alert service bulletin recommends accomplishment of the actions as soon as manpower and materials are available, the FAA has determined that a 90-day compliance time would address the identified unsafe condition in a timely manner. In developing an appropriate compliance time for this AD, the FAA considered not only the manufacturer's recommendation, but the degree of urgency associated with addressing the subject unsafe condition, the average utilization of the affected fleet, and the time necessary to perform the actions. In light of all of these factors, the FAA finds a 90-day compliance time for completion of the actions to be warranted, in that it represents an appropriate interval of time allowable for affected airplanes to continue to operate without compromising safety.

Determination of Rule's Effective Date

Since a situation exists that requires the immediate adoption of this regulation, it is found that notice and opportunity for prior public comment hereon are impracticable, and that good cause exists for making this amendment effective in less than 30 days.

Comments Invited

Although this action is in the form of a final rule that involves requirements affecting flight safety and, thus, was not preceded by notice and an opportunity for public comment, comments are invited on this rule. Interested persons are invited to comment on this rule by submitting such written data, views, or arguments as they may desire.

Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified under the caption **ADDRESSES**. All communications received on or before