

1. A minimum threat of 100 volts rms per meter electric field strength from 10 KHz to 18 GHz.

a. The threat must be applied to the system elements and their associated wiring harnesses without the benefit of airframe shielding.

b. Demonstration of this level of protection is established through system tests and analysis.

2. A threat external to the airframe of the following field strengths for the frequency ranges indicated. Both peak and average field strength components from the Table are to be demonstrated.

Frequency	Field Strength (volts per meter)	
	Peak	Average
10 kHz–100 kHz	50	50
100 kHz–500 kHz	50	50
500 kHz–2 MHz	50	50
2 MHz–30 MHz	100	100
30 MHz–70 MHz	50	50
70 MHz–100 MHz	50	50
100 MHz–200 MHz ...	100	100
200 MHz–400 MHz ...	100	100
400 MHz–700 MHz ...	700	50
700 MHz–1 GHz	700	100
1 GHz–2 GHz	2000	200
2 GHz–4 GHz	3000	200
4 GHz–6 GHz	3000	200
6 GHz–8 GHz	1000	200
8 GHz–12 GHz	3000	300
12 GHz–18 GHz	2000	200
18 GHz–40 GHz	600	200

The field strengths are expressed in terms of peak of the root-mean-square (rms) over the complete modulation period.

The threat levels identified above are the result of an FAA review of existing studies on the subject of HIRF, in light of the ongoing work of the Electromagnetic Effects Harmonization Working Group of the Aviation Rulemaking Advisory Committee.

Applicability

As discussed above, these special conditions are applicable to Learjet Model 55 and 55B airplanes modified by JetCorp. Should JetCorp apply at a later date for a supplemental type certificate to modify any other model included on the same type certificate to incorporate the same novel or unusual design feature, these special conditions would apply to that model as well under the provisions of § 21.101(a)(1).

Conclusion

This action affects only certain novel or unusual design features on Learjet Model 55 and 55B airplanes modified by JetCorp. It is not a rule of general applicability and affects only the applicant who applied to the FAA for approval of these features on the airplane.

The substance of the special conditions for this airplane has been subjected to the notice and comment period in several prior instances and has been derived without substantive change from those previously issued. It is unlikely that prior public comment would result in a significant change from the substance contained herein. For this reason, and because a delay would significantly affect the certification of the airplane, which is imminent, the FAA has determined that prior public notice and comment are unnecessary and impracticable, and good cause exists for adopting these special conditions upon issuance. The FAA is requesting comments to allow interested persons to submit views that may not have been submitted in response to the prior opportunities for comment described above.

List of Subjects in 14 CFR Part 25

Aircraft, Aviation safety, Reporting and recordkeeping requirements.

The authority citation for these special conditions is as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701, 44702, 44704.

The Special Conditions

Accordingly, pursuant to the authority delegated to me by the Administrator, the following special conditions are issued as part of the type certification basis for Learjet Model 55 and 55B airplanes modified by JetCorp.

1. *Protection from Unwanted Effects of High-Intensity Radiated Fields (HIRF).* Each electrical and electronic system that performs critical functions must be designed and installed to ensure that the operation and operational capability of these systems to perform critical functions are not adversely affected when the airplane is exposed to high intensity radiated fields.

2. For the purpose of these special conditions, the following definition applies: *Critical Functions:* Functions whose failure would contribute to or cause a failure condition that would prevent the continued safe flight and landing of the airplane.

Issued in Renton, Washington, on March 7, 2001.

Donald L. Riggin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 01-6372 Filed 3-14-01; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-NE-43-AD; Amendment 39-12143; AD 99-18-18 R1]

RIN 2120-AA64

Airworthiness Directives; Dowty Aerospace Propellers Model R381/6-123-F/5 Propellers

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment revises an existing airworthiness directive (AD) that is applicable to Dowty Aerospace Propellers Model R381/6-123-F/5 propellers, that requires initial and repetitive visual and ultrasonic inspections of propeller blades for cracks across the camber face, and, if blades are found cracked, replacement with serviceable blades. This amendment is prompted by an engineering analysis of field service data and certification testing that indicate that the repetitive visual inspection interval can be safely increased and that the ultrasonic inspections can be eliminated. The actions specified in this proposed AD are intended to detect propeller blade cracks and propagation, which if not detected could result in propeller blade separation and possible aircraft loss of control.

DATES: Effective April 19, 2001. The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of April 19, 2001.

ADDRESSES: The service information referenced in this AD may be obtained from Dowty Aerospace Propellers, Anson Business Park, Cheltenham Road East, Gloucester GL29QN, England; telephone: 44 1452 716000, fax: 44 1452 716001. This information may be examined at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA, or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC. **FOR FURTHER INFORMATION CONTACT:** Kirk Gustafson, Aerospace Engineer, Boston Aircraft Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone: 781-238-7190, fax: 781-238-7199.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) by revising AD 99-18-18, Amendment

39–11284 (64 FR 47661, September 1, 1999), which is applicable to Dowty Aerospace Propellers Model R381/6–123–F/5 propellers, was published in the **Federal Register** on August 21, 2000 (65 FR 50667). The action proposed to increase the propeller blade crack inspection intervals. For repetitive visual inspection intervals, the proposed increase was from 50 to 300 hours time-in-service (TIS) since last inspection, and for repetitive ultrasonic inspection intervals the proposed increase was from 200 to 600 hours TIS.

Comment Received

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comment received.

Eliminate Ultrasonic Inspection and Increase Inspection Interval

A comment from the manufacturer recommends elimination of ultrasonic inspections, based on analysis that concluded that initial and repetitive visual inspection intervals are adequate. The manufacturer states that the engineering analysis of field service data did not reveal a specific root cause for the original cracked blade. It is suspected that an unusual circumstance may have been involved, such as an unreported impact with a ground vehicle. However, to ensure the structural integrity of blades in service, initial and repetitive visual inspections must be done, and, as a result of the analysis, these inspections are being allowed at increased intervals as specified in a new revision to the applicable service bulletin.

The FAA agrees. The engineering data provided to the FAA by the manufacturer indicates there are no specific structural concerns, manufacturing quality issues, or fatigue mechanisms that would justify the need for initial and repetitive ultrasonic inspections, and that an increased repetitive visual inspection interval is appropriate. The inspections were originally proposed by the manufacturer and mandated by the FAA to address an unknown cause for a cracked blade found in service. These inspections were based on a need for a conservative control program as an interim action, while a detailed investigation was performed. As a measure of conservatism, the extended repetitive inspection interval is being retained. The inspection coincides with existing propeller maintenance tasks so as not to create an undue burden while providing additional margin against potential but unanticipated causes for propeller blade

cracks. This amendment has been revised to eliminate the ultrasonic inspections, increase the visual inspection intervals, and reference the newly revised service bulletin.

After careful review of the available data, including the comment noted above, the FAA has determined that air safety and the public interest require the adoption of the rule with the changes described previously. The FAA has determined that these changes will neither increase the economic burden on an operator nor increase the scope of the AD.

Economic Analysis

The FAA estimates that there are six propellers of the affected design installed on aircraft of U.S. registry. The FAA also estimates that it would take approximately four work hours per propeller to accomplish a visual inspection, and that the average labor rate is \$60 per work hour. A propeller will average three visual inspections per year. Based on these figures for the six propellers, the yearly cost impact for this AD is estimated to be \$4,320.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a “significant regulatory action” under Executive Order 12866; (2) is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the

Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by removing Amendment 39–11284 (64 FR 47661, September 1, 1999), and by adding a new airworthiness directive (AD), and by adding a new airworthiness directive (AD), Amendment 39–12143 to read as follows:

99–18–18 R1, Dowty Aerospace Propellers:
Docket 99–NE–43–AD. Revises AD 99–18–18, Amendment 39–11284.

Applicability: Dowty Aerospace Propellers Model R381/6–123–F/5 propellers, installed on but not limited to SAAB 2000 series airplanes.

Note 1: This airworthiness directive (AD) applies to each propeller identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For propellers that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect propeller blade cracks and propagation, which if not detected could result in propeller blade separation and possible aircraft loss of control, accomplish the following:

Visual Inspections

(a) Perform initial and repetitive visual inspections of propeller blades for cracks across the camber face in accordance with the Accomplishment Instructions of Dowty Aerospace Propellers Service Bulletin (SB) No. S2000–61–75, Revision 4, dated September 28, 2000, as follows:

(1) Initially, conduct a visual inspection within 50 hours time-in-service (TIS) after the effective date of this AD.

(2) Thereafter, inspect at intervals not to exceed 600 hours TIS since last inspection.

(3) Replace cracked propeller blades prior to further flight with serviceable blades.

(b) [Reserved]

Alternative Method of Compliance

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be

used if approved by the Manager, Boston Aircraft Certification Office (ACO). Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Boston ACO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this airworthiness directive, if any, may be obtained from the Boston ACO.

Special Flight Permits

(d) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the aircraft to a location where the inspection requirements of this AD can be accomplished.

Incorporation by Reference

(e) The actions required by this AD must be done in accordance with the Accomplishment Instructions of Dowty Aerospace Propellers Service Bulletin (SB) No. S2000-61-75, Revision 4, dated September 28, 2000. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Dowty Aerospace Propellers, Anson Business Park, Cheltenham Road East, Gloucester GL29QN, England; telephone: 44 1452 716000, fax: 44 1452 716001. Copies may be inspected at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA, or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

Effective Date

(f) This amendment becomes effective on April 19, 2001.

Issued in Burlington, Massachusetts, on March 1, 2001.

David A. Downey,

Acting Manager, Engine and Propeller Directorate, Aircraft Certification Service
[FR Doc. 01-5735 Filed 3-14-01; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2000-NE-43-AD; Amendment 39-12144; AD 2001-05-07]

RIN 2120-AA64

Airworthiness Directives; Pratt & Whitney PW4000 Series Turbofan Engines

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule, request for comments.

SUMMARY: This amendment supersedes airworthiness directive (AD) 2000-25-06, dated December 5, 2000, that is applicable to certain Pratt & Whitney (PW) PW4000 turbofan engines with the current design low pressure turbine (LPT) 4th stage air seal installed. That AD currently requires, based on engine model, replacement of the current design seal with a new design seal, or with a modified seal. This amendment adds the listing of certain engine serial numbers, to correct an error in the applicability section of AD 2000-25-06, for engines affected by PW Service Bulletin (SB) PW4 ENG 72-657, Revision 1, dated July, 19, 2000. This correction is prompted by comments received on AD 2000-25-06. The actions specified by this AD are intended to reduce stresses that could lead to LPT 4th stage air seal cracking, resulting in seal fracture, uncontained engine failure, and damage to the airplane.

DATES: Effective date March 30, 2001.

Comments for inclusion in the rules docket must be received on or before May 14, 2001.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), New England Region, Office of the Regional Counsel, Attention: Rules Docket No. 2000-NE-43-AD, 12 New England Executive Park, Burlington, MA 01803-5299. Comments may also be sent via the Internet using the following address: "9-ane-adcomment@faa.gov". Comments sent via the Internet must contain the docket number in the subject line.

The service information referenced in this AD may be obtained from Pratt & Whitney, 400 Main St., East Hartford, CT 06108; telephone: 860 565-6600, fax: 860 565-4503. This information may be examined at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA.

FOR FURTHER INFORMATION CONTACT: Tara Goodman, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone: 781 238-7130; fax: 781 238-7199.

SUPPLEMENTARY INFORMATION: On December 5, 2000, the FAA issued AD 2000-25-06, Amendment 39-12040, (65 FR 78083) dated December 14, 2000 that is applicable to certain Pratt & Whitney (PW) PW4000 turbofan engines. That AD requires replacement of the current design LPT 4th stage air seal with a new design seal, or with a modified seal. That action was prompted by reports of cracks in LPT 4th stage air seals. That

condition, if not corrected, could lead to LPT 4th stage air seal cracking, resulting in seal fracture, uncontained engine failure, and damage to the airplane.

Since the issuance of that AD, comments were received on AD 2000-25-06, stating that an error exists in Table 1 which incorrectly includes a limited population of engines affected by PW SB 72-657, Revision 1, dated July 19, 2000. The FAA agrees that an error was inadvertently made, and that the need to correct Table 1 warrants a new superseding final rule, request for comments, to address those comments and other comments received.

Since an unsafe condition has been identified that is likely to exist or develop on other PW4000 turbofan engines of the same type design, this AD supersedes AD 2000-25-06 to require the correction of engine populations affected.

Comments Received

Interested persons have been afforded an opportunity to comment on the Final Rule, Request for Comments, AD 2000-25-06. Due consideration has been given to the comments received, and as a result, this superseding final rule, request for comments AD is deemed necessary.

Table 1 Error

Six commenters state that an error exists in Table 1, that includes a limited population of engines affected by PW SB 72-657, Revision 1, dated July 19, 2000.

The FAA agrees. The error was made inadvertently. This amendment corrects that error by listing certain engine serial numbers in a table to clarify applicability for engines affected by PW Service Bulletin (SB) PW4 ENG 72-657, Revision 1, dated July, 19, 2000.

Concern for Future AD Revision or AMOC

One commenter states a concern that with regard to Table 2, future air seal designs will warrant an AD revision or an alternative method of compliance (AMOC). The commenter requests that this amendment: (1) Allow for future air seal part numbers (P/N's), (2) revise Table 1 accordingly, and (3) eliminate Table 2.

The FAA does not agree. This AD is applicable to engines with LPT 4th stage air seals P/N 50N478 or 50N478-001 installed and requires a one-time replacement of the air seal, according to Table 2. There is no on-going requirement to use only the parts listed in Table 2 in the future. Table 2 specifies what is a serviceable part. For PW4000 100-inch models, the relevant