

**Figure 5**

For McDonnell Douglas Model DC-9-11, DC-9-12, DC-9-13, DC-9-14, DC-9-15, DC-9-15F, DC-9-21, DC-9-31, DC-9-32, DC-9-32 (VC-9C), DC-9-32F, DC-9-32F (C-9A, C-9B), DC-9-33F, DC-9-34, DC-9-34F, DC-9-41, and DC-9-51 Airplanes:

Insert the information in this figure into the "Emergency Procedures" section of the FAA-approved Airplane Flight Manual.

**"CABIN ALTITUDE WARNING/RAPID DEPRESSURIZATION/EMERGENCY DESCENT"**

**Phase I and II**

If a cabin altitude warning occurs:

|                                |                                  |
|--------------------------------|----------------------------------|
| Crew Oxygen Masks              | ON/100%                          |
| Manual Pressurization Control. | FULL FORWARD AND MANUALLY LOCKED |

**Note:** Manual Pressurization control forces may be high, apply forces as required."

The rest of the steps under this heading in the AFM are unchanged.

**Figure 6**

For McDonnell Douglas Model DC-9-81 (MD-81), DC-9-82 (MD-82), DC-9-83 (MD-83), DC-9-87 (MD-87), and MD-88 Airplanes:

Insert the information in this figure into the "Emergency Procedures" section of the FAA-approved Airplane Flight Manual.

**"CABIN ALTITUDE WARNING/RAPID DEPRESSURIZATION/EMERGENCY DESCENT"**

**Phase I and II**

If the cabin altitude warning occurs:

|                                |                                  |
|--------------------------------|----------------------------------|
| Crew Oxygen Mask ..            | ON/100%                          |
| Manual Pressurization Control. | FULL FORWARD AND MANUALLY LOCKED |

**Note:** Manual Pressurization control forces may be high, apply forces as required."

The rest of the steps under this heading in the AFM are unchanged.

**Figure 7**

For McDonnell Douglas MD-90-30 Airplanes:

Insert the information in this figure into the "Emergency Procedures" section of the FAA-approved Airplane Flight Manual.

**"CABIN ALTITUDE WARNING OR RAPID DEPRESSURIZATION"**

If the cabin altitude warning occurs:

OXY MASKS ..... ON/100%"

The rest of the steps under this heading in the AFM are unchanged.

**Figure 8**

For McDonnell Douglas Model DC-10-10, DC-10-10F, DC-10-15, DC-10-30, DC-10-30F, DC-10-30F (KC-10A, KDC-10), DC-10-40, and DC-10-40F Airplanes:

Insert the information in this figure into the "Emergency Procedures" section of the FAA-approved Airplane Flight Manual.

**"CABIN ALTITUDE WARNING OR RAPID DEPRESSURIZATION/EMERGENCY DESCENT"**

**Recall**

If the cabin altitude warning occurs:

|                    |                                                               |
|--------------------|---------------------------------------------------------------|
| Oxygen Masks ..... | ON/100%                                                       |
| Cabin              |                                                               |
| OUTFLOW VALVE ..   | VERIFY CLOSED (CLOSE ELECTRICALLY OR MANUALLY IF NOT CLOSED)" |

The rest of the steps under this heading in the AFM are unchanged.

**Figure 9**

For McDonnell Douglas Model MD-10-10F, MD-10-30F, MD-11, and MD-11F Airplanes:

Insert the information in this figure into the "Emergency Procedures" section of the FAA-approved Airplane Flight Manual.

**"CABIN ALTITUDE WARNING OR CABIN ALTITUDE"**

If the cabin altitude warning occurs:

|                     |                |
|---------------------|----------------|
|                     | Memory Item    |
| Oxygen Masks .....  | ON/100%        |
| Outflow Valve ..... | Verify Closed" |

The rest of the steps under this heading in the AFM are unchanged.

**Alternative Methods of Compliance**

(b) In accordance with 14 CFR 39.19, the Manager, Seattle Aircraft Certification Office (ACO), FAA, or the Manager, Los Angeles ACO, FAA, is authorized to approve alternative methods of compliance for this AD.

Issued in Renton, Washington, on July 1, 2003.

**Vi L. Lipski,**

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-17317 Filed 7-8-03; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

[Docket No. 2002-NM-207-AD]

**RIN 2120-AA64**

**Airworthiness Directives; Boeing Model 747-100, -100B, -100B SUD, -200B, -200C, -200F, -300, 747SR and 747SP Series Airplanes Equipped With Pratt & Whitney JT9D Series Engines**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Boeing transport category airplanes listed above. This proposal

would require drilling witness holes through the cowl skin at the cowl latch locations in the left-hand side of the cowl panel assembly of each engine. This action is necessary to prevent improper connection of the latch, which could result in separation of a cowl panel from the airplane. Such separation could cause damage to the airplane, consequent rapid depressurization, and hazards to persons or property on the ground. This action is intended to address the identified unsafe condition.

**DATES:** Comments must be received by August 25, 2003.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2002-NM-207-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: [9-anm-nprmcomment@faa.gov](mailto:9-anm-nprmcomment@faa.gov). Comments sent via fax or the Internet must contain "Docket No. 2002-NM-207-AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 or 2000 or ASCII text.

The service information referenced in the proposed rule may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington.

**FOR FURTHER INFORMATION CONTACT:** Dan Kinney, Aerospace Engineer, Propulsion Branch, ANM-140S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 917-6499; fax (425) 917-6590.

**SUPPLEMENTARY INFORMATION:****Comments Invited**

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained

in this action may be changed in light of the comments received.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.
- For each issue, state what specific change to the proposed AD is being requested.
- Include justification (e.g., reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2002-NM-207-AD." The postcard will be date stamped and returned to the commenter.

#### **Availability of NPRMs**

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2002-NM-207-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

#### **Discussion**

The FAA has received a report of in-flight separation of the cowl panels on the left- and right-hand sides of a Model 747 series airplane. Investigation revealed that the probable cause of the separation was improper latching of the cowl latch assemblies. The original cowl latch design allowed the latch to appear connected and latched, when it may not have been attached to the mating U-bolt. An improved design was developed for the latch to prevent incorrect latching, but in-service experience has shown that improper latching is still possible, even with the new latch. The addition of witness holes will allow the mechanic to visually inspect before a flight to determine that the cowl latch is properly connected. Improper connection of the latch could result in separation of a cowl panel from the airplane. Such separation could cause damage to the airplane, consequent

rapid depressurization, and hazards to persons or property on the ground.

#### **Explanation of Relevant Service Information**

The FAA has reviewed and approved Boeing Service Bulletin 747-71-2301, dated May 30, 2002, which describes procedures for drilling 0.50-inch-diameter witness holes through the cowl skin at each of the six cowl latch locations located on the left-hand side of the cowl panel assembly of each engine. Accomplishment of the actions specified in the service bulletin is intended to adequately address the identified unsafe condition.

#### **Explanation of Requirements of Proposed Rule**

Since an unsafe condition has been identified that is likely to exist or develop on other products of this same type design, the proposed AD would require accomplishment of the actions specified in the service bulletin described previously, except as discussed below.

#### **Explanation of Compliance Time**

We have determined that a compliance time of 36 months for initiating the proposed actions is warranted, in that it represents an appropriate interval of time allowable for affected airplanes to continue to operate without compromising safety.

#### **Changes to 14 CFR Part 39/Effect on the Proposed AD**

On July 10, 2002, the FAA issued a new version of 14 CFR part 39 (67 FR 47997, July 22, 2002), which governs the FAA's airworthiness directives system. The regulation now includes material that relates to altered products, special flight permits, and alternative methods of compliance. Because we have now included this material in part 39, we no longer need to include it in each individual AD; however, this AD identifies the office authorized to approve alternative methods of compliance.

#### **Explanation of Cost Increase**

We have reviewed the figures we have used over the past several years to calculate AD costs to operators. To account for various inflationary costs in the airline industry, we find it necessary to increase the labor rate used in these calculations from \$60 per work hour to \$65 per work hour. The cost impact information, below, has been revised to reflect this increase in the specified hourly labor rate.

#### **Cost Impact**

There are approximately 345 airplanes of the affected design in the worldwide fleet. The FAA estimates that 108 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 8 work hours per airplane (2 work hours per engine) to accomplish the proposed actions, and that the average labor rate is \$65 per work hour. Based on these figures, the cost impact of the proposed AD on U.S. operators is estimated to be \$56,160, or \$520 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this proposed AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

#### **Regulatory Impact**

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

#### **List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Safety.

#### **The Proposed Amendment**

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation

Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

### **PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### **§ 39.13 [Amended]**

2. Section 39.13 is amended by adding the following new airworthiness directive:

**Boeing:** Docket 2002–NM–207–AD.

Applicability: Model 747–100, –100B, –100B SUD, –200B, –200C, –200F, –300, 747SR, and 747SP series airplanes; equipped with Pratt & Whitney JT9D series engines; line numbers 1 through 669 inclusive; certificated in any category.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent improper connection of the cowl latch located in the left-hand side of the cowl panel assembly of each engine, which could result in separation of a cowl panel from the airplane, accomplish the following:

#### **Drill Holes**

(a) Within 36 months after the effective date of this AD: Drill witness holes through the cowl skin at each of the six cowl latch locations located on the left-hand side of the cowl panel assembly of each engine, per paragraphs 3.B.1. through 3.B.4. of the Accomplishment Instructions of Boeing Service Bulletin 747–71–2301, dated May 30, 2002.

#### **Alternative Methods of Compliance**

(b) In accordance with 14 CFR 39.19, the Manager, Seattle Aircraft Certification Office, FAA, is authorized to approve alternative methods of compliance for this AD.

Issued in Renton, Washington, on July 1, 2003.

**Vi L. Lipski,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 03–17318 Filed 7–8–03; 8:45 am]

**BILLING CODE 4910–13–P**

## **DEPARTMENT OF TRANSPORTATION**

### **Federal Aviation Administration**

#### **14 CFR Part 39**

[Docket No. 2003–NM–143–AD]

RIN 2120–AA64

#### **Airworthiness Directives; Bombardier Model CL–600–2B19 (Regional Jet Series 100 & 440) Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Model CL–600–2B19 (Regional Jet Series 100 & 440) airplanes. This proposal would require revising the airworthiness limitations section of the Instructions for Continued Airworthiness by incorporating new structural inspection intervals for the vertical beams of the pressure bulkheads at fuselage stations 409+128 and 559; repairing the vertical beams if necessary; and submitting inspection findings to the airplane manufacturer. This action is necessary to detect and correct, in a timely manner, fatigue cracks in the vertical beams of the pressure bulkheads at fuselage stations 409+128 and 559, which could result in the reduced structural integrity of the airplane. This action is intended to address the identified unsafe condition.

**DATES:** Comments must be received by August 8, 2003.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–114, Attention: Rules Docket No. 2003–NM–143–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227–1232. Comments may also be sent via the Internet using the following address: [9-anm-nprmcomment@faa.gov](mailto:9-anm-nprmcomment@faa.gov). Comments sent via fax or the Internet must contain “Docket No. 2003–NM–143–AD” in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 or 2000 or ASCII text.

The service information referenced in the proposed rule may be obtained from Bombardier, Inc., Canadair, Aerospace Group, P.O. Box 6087, Station Centre-ville, Montreal, Quebec H3C 3G9, Canada. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York.

**FOR FURTHER INFORMATION CONTACT:** Serge Napoleon, Aerospace Engineer, Airframe and Propulsion Branch, ANE–171, FAA, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York 11581; telephone (516) 256–7512; fax (516) 568–2716.

**SUPPLEMENTARY INFORMATION:**

### **Comments Invited**

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this action may be changed in light of the comments received.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.
- For each issue, state what specific change to the proposed AD is being requested.
- Include justification (e.g., reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: “Comments to Docket Number 2003–NM–143–AD.” The postcard will be date stamped and returned to the commenter.

### **Availability of NPRMs**

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM–114, Attention: Rules Docket No. 2003–NM–143–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056.

### **Discussion**

Transport Canada Civil Aviation (TCCA), which is the airworthiness authority for Canada, notified the FAA that an unsafe condition may exist on certain Bombardier Model CL–600–2B19 (Regional Jet series 100 & 440) airplanes. TCCA advises that fatigue cracks were found in the vertical beams of the pressure bulkheads at fuselage stations 409+128 and 559. This