

Actions	Compliance	Procedures
(1) Inspect the rudder control lever shafts, part numbers (P/N) 2/N-45-1102, 1/N-45-1103, and 1/N-45-1104 (or FAA-approved equivalent part numbers) for cracks. Use dye penetrant while the shaft is installed; or either dye penetrant inspection or magnetic particle methods if the shaft is removed. (2) Inspect all lever shaft side plates on P/Ns 2/N-45-1102, 1/N-45-1103, and 1/N-45-1104 (or FAA-approved equivalent part numbers) by measuring the thickness for discrepancies. (3) Visually inspect all rudder control lever shafts P/Ns 2/N-45-1102, 1/N-45-1103, and 1/N-45-1104 (or FAA-approved equivalent part numbers) for cracks. (4) If damage is found during any inspection required by this AD. (i) For lever shafts found with crack damage, replace with new or serviceable items (ii) For discrepancies in the thickness of lever shaft side plates, obtain a repair scheme from the manufacturer through FAA at the address specified in paragraph (e) of this AD and incorporate this repair scheme (iii) Repairable and nonrepairable damage is defined in the service information	Within the next 100 hours time-in-service (TIS) after September 8, 2003 (the effective date of this AD), unless already accomplished. Within the next 100 hours TIS after September 8, 2003 (the effective date of this AD), unless already accomplished. Repetitively inspect at intervals not to exceed 100 hours TIS after the inspection required in paragraph (d)(1) of this AD. Prior to further flight after any inspection required by this AD.	In accordance with Nomad Alert Service Bulletin ANMD-27-51, dated September 13, 2002, and the applicable maintenance manual. In accordance with Nomad Alert Service Bulletin ANMD-27-51, dated September 13, 2002, and the applicable maintenance manual. In accordance with Nomad Alert Service Bulletin ANMD-27-51, dated September 13, 2002, and the applicable maintenance manual. In accordance with Nomad Alert Service Bulletin ANMD-27-51, dated September 13, 2002, and the applicable maintenance manual.

(e) *Can I comply with this AD in any other way?*

(1) To use an alternative method of compliance or adjust the compliance time, follow the procedures in 14 CFR 39.13. Send these requests to the Manager, Los Angeles Aircraft Certification Office. For information on any already approved alternative methods of compliance, contact Ron Atmur, Aerospace Engineer, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712; telephone (562) 627-5224; facsimile (562) 627-5210.

(2) Alternative methods of compliance approved in accordance with AD 82-12-06, which is superseded by this AD, are not approved as alternative methods of compliance with this AD.

(f) *Are any service bulletins incorporated into this AD by reference?* Actions required by this AD must be done in accordance with Nomad Alert Service Bulletin ANMD-27-51, dated September 13, 2002. The Director of the Federal Register approved this incorporation by reference under 5 U.S.C. 552(a) and 1 CFR part 51. You may get copies from Nomad Operations, Aerospace Support Division, Boeing Australia, PO Box 767, Brisbane, QLD 4000 Australia; telephone 61 7 3306 3366; facsimile 61 7 3306 3111. You may view copies at the FAA, Central Region, Office of the Regional Counsel, 901 Locust, Room 506, Kansas City, Missouri, or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(g) *Does this AD action affect any existing AD actions?* This amendment supersedes AD 82-12-06, Amendment 39-4399.

Note: The subject of this AD is addressed in Australian AD GAF-N22/44, dated November 14, 2002.

(h) *When does this amendment become effective?* This amendment becomes effective on September 8, 2003.

Issued in Kansas City, Missouri, on July 10, 2003.

Dorenda D. Baker,

Acting Manager, Small Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-17945 Filed 7-18-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2003-NM-165-AD; Amendment 39-13225; AD 2003-14-06]

RIN 2120-AA64

Airworthiness Directives; Boeing Model 737-200, -200C, -300, -400, and -500 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; correction.

SUMMARY: This document corrects a typographical error that appeared in airworthiness directive (AD) 2003-14-06 that was published in the **Federal Register** on July 9, 2003 (68 FR 40759). The typographical error resulted in an incorrect AD number in one location of the document. This AD is applicable to certain Boeing Model 737-200, -200C, -300, -400, and -500 series airplanes. This AD requires repetitive inspections

for cracking of certain lap splices, and corrective action if necessary.

DATES: Effective July 14, 2003.

FOR FURTHER INFORMATION CONTACT:

Duong Tran, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 917-6452; fax (425) 917-6590.

SUPPLEMENTARY INFORMATION:

Airworthiness Directive (AD) 2003-14-06, amendment 39-13225, applicable to certain Boeing Model 737-200, -200C, -300, -400, and -500 series airplanes, was published in the **Federal Register** on July 9, 2003 (68 FR 40759). That AD requires repetitive inspections for cracking of certain lap splices, and corrective action if necessary.

As published, the AD number appears as "2003-14-60" in the Product Identification line in the regulatory text of the AD. The correct AD number is 2003-14-06. The number is referenced correctly throughout the remainder of the document.

Since no other part of the regulatory information has been changed, the final rule is not being republished in the **Federal Register**.

The effective date of this AD remains July 14, 2003.

§ 39.13 [Corrected]

■ On page 40761, in the first column, the Product Identification line of AD 2003-14-06 is corrected to read as follows:

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2003-14-06 Boeing: Amendment 39-13225.
Docket 2003-NM-165-AD.

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Issued in Renton, Washington, on July 15, 2003.

Ali Bahrami,

*Acting Manager, Transport Airplane
Directorate, Aircraft Certification Service.*

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-SW-56-AD; Amendment 39-13231; AD 2003-14-12]

RIN 2120-AA64

Airworthiness Directives; Eurocopter France Model SA330F, G, and J; AS332C, L, and L1; SA341G; SA342J; AS350B, BA, B1, B2, B3, and D; AS355E, F, F1, F2 and N; SA-365C, C1, and C2; SA-365N and N1; and AS-365N2 and N3 Helicopters

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD) for the specified Eurocopter France (ECF) model helicopters that requires determining whether specified main or tail rotor (rotor) parts are installed and, if so, updating and recording the correct hours time-in-service (TIS) or cycles of each part. If the hours TIS or cycles of any rotor part exceed its life limit, this AD also requires replacing that part with an airworthy part within 50 hours TIS. This amendment is prompted by the need to correct the Equipment Log Card (FME) to accurately reflect the total hours TIS and cycles of certain repaired or overhauled rotor parts. The actions specified by this AD are intended to prevent failure of a life limited rotor part, loss of a rotor, and subsequent loss of control of the helicopter.

DATES: Effective August 25, 2003.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of August 25, 2003.

ADDRESSES: The service information referenced in this AD may be obtained from American Eurocopter Corporation, 2701 Forum Drive, Grand Prairie, Texas 75053-4005, telephone (972) 641-3460, fax (972) 641-3527. This information may be examined at the FAA, Office of the Regional Counsel, Southwest

Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Uday Garadi, Aviation Safety Engineer, FAA, Rotorcraft Directorate, Rotorcraft Standards Staff, Fort Worth, Texas 76193-0110, telephone (817) 222-5123, fax (817) 222-5961.

SUPPLEMENTARY INFORMATION:

A proposal to amend 14 CFR part 39 to include an AD for the specified model helicopters was published in the **Federal Register** on March 14, 2003 (68 FR 12318). That action proposed to require determining whether specified rotor parts are installed and, if so, updating and recording the correct hours TIS or cycles of each part. If the hours TIS or cycles of any rotor part exceed its life limit, this AD would also require replacing that part with an airworthy part within 50 hours TIS.

The Direction Generale De L'Aviation Civile (DGAC), the airworthiness authority for France, notified the FAA that an unsafe condition may exist on ECF Model SA330F, G, and J; AS332C, L, and L1; SA341G; SA342J; AS350B, BA, B1, B2, B3, and D; AS355E, F, F1, F2 and N; SA-365C, C1, and C2; SA-365N and N1; and AS-365N2 and N3 helicopters. The DGAC advises of the discovery of a discrepancy in the computer program used to carry over the number of operating hours of parts following repair or overhaul, which is the cause of incorrect completion of FMEs.

ECF has issued the following Alert Telexes for the helicopter model series specified: Nos. 65.110 for SA330, 62.00.58 for AS332, 65.60 for SA341 and SA342, 62.00.25 for AS350, 62.00.27 for AS355, 65.41 for SA-365C, and 62.00.19 for AS-365N, all dated August 13, 2002. These alert telexes specify correcting the FME to list the correct total number of operating hours and cycles for specified parts installed on dynamic components. After correcting the FME, if the parts have exceeded their life limit, the alert telexes specify removing or monitoring the parts. The DGAC classified these alert telexes as mandatory and issued AD No. 2002-452(A), dated September 4, 2002, to ensure the continued airworthiness of these helicopters in France.

These helicopter models are manufactured in France and are type certificated for operation in the United States under the provisions of 14 CFR 21.29 and the applicable bilateral agreement. Pursuant to the applicable

bilateral agreement, the DGAC has kept the FAA informed of the situation described above. The FAA has examined the findings of the DGAC, reviewed all available information, and determined that AD action is necessary for products of these type designs that are certificated for operation in the United States.

This unsafe condition is likely to exist or develop on other helicopters of the same type designs registered in the United States. Therefore, the AD requires, within 10 hours TIS, determining whether the specified rotor part and serial numbers are installed by reference to the FME and, if installed, correcting the hours TIS and cycles. If a part exceeds its life limit, the AD requires replacing the part within 50 hours TIS. The actions would be required for the parts listed in the appendix of the alert telexes described previously.

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were received on the proposal or the FAA's determination of the cost to the public. The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

On July 10, 2002, the FAA issued a new version of 14 CFR part 39 (67 FR 47997, July 22, 2002), which governs the FAA's AD system. The regulation now includes material that relates to altered products, special flight permits, and alternative methods of compliance. However, for clarity and consistency in this final rule, we have retained the language of the NPRM regarding that material.

The FAA estimates that this AD will affect 760 helicopters of U.S. registry, and the required actions will take approximately 1 work hour to determine the part and serial number and 8 hours to replace each affected part on 38 helicopters (5 percent of the total affected helicopters). The average labor rate is \$60 per work hour. Required parts will cost approximately \$64,560 depending on which part will be replaced. Based on these figures, we estimate the total cost impact of the AD on U.S. operators will be \$2,517,120.

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.