

action for the repetitive tests required by paragraph (a) of this AD.

(1) Install provisional wiring for the additional locking system on the thrust reversers, in accordance with the Accomplishment Instructions of Boeing Service Bulletin 747-78-2134, Revision 3, dated March 19, 1998.

(2) Remove the thrust reverser sequencing mechanism and install a solenoid-operated shutoff valve in accordance with Boeing Service Bulletin 747-78-2052, Revision 5, dated February 22, 1996.

(3) Install an additional locking system on each thrust reverser in accordance with the Accomplishment Instructions of Boeing Service Bulletin 747-78-2152, Revision 5, dated June 14, 2001; or Revision 6, dated October 24, 2002.

Repetitive Tests

(d) Within 3,000 flight hours after accomplishment of paragraph (c) of this AD: Perform a functional test to detect discrepancies of the additional locking system on each thrust reverser in accordance with the procedures described in the Boeing 747 Airplane Maintenance Manual (AMM), Section 78-34-11, dated October 25, 1997. Prior to further flight, correct any discrepancy detected and repeat the functional test of that repair in accordance with the procedures described in the AMM. Repeat the functional tests thereafter at intervals not to exceed 3,000 flight hours.

Dispatch Limitations

(e) If, after incorporation of the modification required by paragraph (c)(3) of this AD on any airplane, it becomes necessary to install a thrust reverser assembly that does not have the additional locking system installed, dispatch of the airplane is allowed in accordance with the provisions and limitations specified in the operator's FAA-approved Master Minimum Equipment List, provided that the thrust reverser assembly that does not have the additional locking system installed is deactivated in accordance with Item 78-1, Section 2, of Boeing Document D6-33391, "Boeing 747-100/-200/-300/SP Dispatch Deviations Procedures Guide," Revision 25, dated July 26, 2002. No more than one thrust reverser on any airplane may be deactivated under the provisions of this paragraph. Within 10 days after deactivation of the thrust reverser, install a thrust reverser assembly that has the additional locking system installed and reactivate the thrust reverser.

(f) If, prior to incorporation of the modification required by paragraph (c)(3) of this AD on any airplane, it becomes necessary to install a thrust reverser assembly that has the additional locking system installed, dispatch of the airplane is allowed in accordance with the provisions and limitations specified in the operator's FAA-approved Master Minimum Equipment List, provided that the thrust reverser assembly that has the additional locking system installed is deactivated in accordance with Item 78-1, Section 2, of Boeing Document D6-33391, "Boeing 747-100/-200/-300/SP Dispatch Deviations Procedures Guide," Revision 25, dated July 26, 2002. No more

than one thrust reverser on any airplane may be deactivated under the provisions of this paragraph. Within 10 days after deactivation of the thrust reverser, install a thrust reverser assembly that does not have the additional locking system installed and reactivate the thrust reverser.

Alternative Methods of Compliance

(g)(1) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

(2) Alternative methods of compliance, approved previously in accordance with paragraphs (a) and (b) of AD 95-16-02, amendment 39-9321, are approved as alternative methods of compliance with the corresponding paragraphs in this AD.

Note 4: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permits

(h) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(i) Unless otherwise specified in this AD, the actions shall be done in accordance with Boeing Service Bulletin 747-78-2134, Revision 3, dated March 19, 1998; Boeing Service Bulletin 747-78-2052, Revision 5, dated February 22, 1996; Boeing Service Bulletin 747-78-2152, Revision 5, dated June 14, 2001; Boeing Service Bulletin 747-78-2152, Revision 6, dated October 24, 2002; and Boeing Alert Service Bulletin 747-78A2131, dated September 15, 1994; as applicable.

(1) This incorporation by reference of Boeing Service Bulletin 747-78-2134, Revision 3, dated March 19, 1998; Boeing Service Bulletin 747-78-2052, Revision 5, dated February 22, 1996; Boeing Service Bulletin 747-78-2152, Revision 5, dated June 14, 2001; and Boeing Service Bulletin 747-78-2152, Revision 6, dated October 24, 2002; is approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51.

(2) The incorporation by reference of Boeing Alert Service Bulletin 747-78A2131, dated September 15, 1994, was approved previously by the Director of the Federal Register as of September 5, 1995 (60 FR 39631, August 3, 1995).

(3) Copies may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Effective Date

(j) This amendment becomes effective on November 18, 2003.

Issued in Renton, Washington, on October 3, 2003.

Ali Bahrami,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03-25700 Filed 10-10-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-NE-21-AD; Amendment 39-13337; AD 2003-05-10R1]

RIN 2120-AA64

Airworthiness Directives; General Electric Company CF34-3A1, -3B, and -3B1 Turbofan Engines

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment revises an existing airworthiness directive (AD), that applies to General Electric Company (GE) CF34-3A1, -3B, and -3B1 turbofan engines with scavenge screens part numbers (P/Ns) 4047T95P01 and 5054T86G02 installed in the B-sump oil scavenge system. That AD currently requires initial and repetitive visual inspections and cleaning of the B-sump scavenge screens until a screenless fitting is installed. This amendment requires the same initial and repetitive visual inspections and cleaning of the B-sump scavenge screens until a screenless fitting is installed. This amendment also corrects a typographical error, and introduces a less restrictive terminating action schedule. This amendment is prompted by the need to correct a typographical error and by the need to introduce a less restrictive terminating action schedule. We are issuing this AD to prevent B-sump scavenge screen blockage due to coking which could result in ignition of B-sump oil in the secondary air system, fan drive shaft separation, and uncontained engine failure.

DATES: Effective November 18, 2003. The incorporation by reference of certain publications, listed in the regulations, was approved previously by the Director of the Federal Register as of April 2, 2003 (68 FR 12806; March 18, 2003).

ADDRESSES: The service information referenced in this AD may be obtained from GE Aircraft Engines, 1000 Western

Avenue, Lynn, MA 01910; Attention: CF34 Product Support Engineering, Mail Zone: 34017; telephone (781) 594-6323; fax (781) 594-0600. This information may be examined, by appointment, at the Federal Aviation Administration (FAA), New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Barbara Caufield, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone (781) 238-7146; fax (781) 238-7199.

SUPPLEMENTARY INFORMATION:

A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) by revising AD 2003-05-10, Amendment 39-13086 (68 FR 12806, March 18, 2003), which applies to GE CF34-3A1, -3B, and -3B1 turbofan engines with scavenge screens P/Ns 4047T95P01 and 5054T86G02 installed in the B-sump oil scavenge system was published in the **Federal Register** on July 8, 2003 (68 FR 40573). That action proposed to require the same initial and repetitive visual inspections and cleaning of the B-sump scavenge screens until a screenless fitting is installed as in AD 2003-05-10. That action also proposed to correct a typographical error, and introduces a less restrictive terminating action schedule in accordance with GE Alert Service Bulletin (ASB) CF34-AL S/B 79-A0014, Revision 3, dated January 31, 2003; ASB CF34-BJ S/B 79-A0015, Revision 3, dated January 31, 2003; GE ASB CF34-AL S/B 79-A0016 and ASB CF34-BJ S/B 79-A0017, both dated June 17, 2002.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were received on the proposal or the FAA's determination of the cost to the public. The FAA has determined that air safety and the

public interest require the adoption of the rule as proposed.

Regulatory Analysis

This final rule does not have federalism implications, as defined in Executive Order 13132, because it would not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Accordingly, the FAA has not consulted with state authorities prior to publication of this final rule.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. Section 39.13 is amended by removing Amendment 39-13086 (68 FR

12806, March 18, 2003) and by adding a new airworthiness directive, Amendment 39-13337, to read as follows:

2003-05-10R1 General Electric Company:
Amendment 39-13337. Docket No. 2001-NE-21-AD. Revises AD 2003-05-10, Amendment 39-13086.

Applicability

This airworthiness directive (AD) applies to General Electric Company (GE) CF34-3A1, -3B, and -3B1 turbofan engines with scavenge screens part numbers (P/Ns) 4047T95P01 and 5054T86G02 installed in the B-sump oil scavenge system. These engines are installed on, but not limited to, Bombardier Inc. (Canadair) Model CL-600-2A12, CL-600-2B16, and CL-600-2B19 airplanes.

Note 1: This AD applies to each engine identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For engines that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance

Compliance with this AD is required as indicated, unless already done.

To prevent B-sump scavenge screen blockage due to coking, which could result in ignition of B-sump oil in the secondary air system, fan drive shaft separation, and uncontained engine failure, do the following:

Initial Inspection and Cleaning of B-Sump Screens

(a) Perform an initial visual inspection and cleaning of scavenge screens, P/Ns 4047T95P01 and 5054T86G02, installed in the B-sump oil scavenge system, in accordance with Paragraphs 3A through 3B of the Accomplishment Instructions of GE Aircraft Engines (GE) Alert Service Bulletin (ASB) CF34-AL S/B 79-A0014, Revision 3, dated January 31, 2003; or ASB CF34-BJ S/B 79-A0015, Revision 3, dated January 31, 2003; and the following table:

INITIAL INSPECTION AND CLEANING SCHEDULE

Engine hours time-since-new (TSN) or time-since-last-shop-visit (TSLSV)	Inspect and clean
(1) Fewer than 4,000 hours TSN or fewer than 4,000 hours TSLSV if it can be confirmed that both the B-sump scavenge screens were cleaned and the B-sump and combustor frame (strut tubes) were removed from the engine and cleaned at that prior shop visit.	Before 4,000 hours TSN or TSLSV.
(2) Fewer than 1,000 hours TSLSV if it can NOT be confirmed that both the B-sump scavenge screens were cleaned and the B-sump and combustor frame (strut tubes) were removed from the engine and cleaned at that prior shop visit.	Before 1,000 hours TSLSV.

INITIAL INSPECTION AND CLEANING SCHEDULE—Continued

Engine hours time-since-new (TSN) or time-since-last-shop-visit (TSLSV)	Inspect and clean
(3) 4,000 hours or greater TSN or 4,000 hours or greater TSLSV if it can be confirmed that both the B-sump scavenge screens were cleaned and the B-sump and combustor frame (strut tubes) were removed from the engine and cleaned at that prior shop visit, or 1,000 hours or greater TSLSV if it can NOT be confirmed that both the B-sump and combustor frame (strut tubes) were removed from the engine and cleaned at that prior shop visit.	Within 500 hours time-in-service (TIS) after the effective date of this AD.

Repetitive Inspections and Cleaning

(b) Perform repetitive visual inspections and cleaning of scavenge screens, P/Ns 4047T95P01 and 5054T86G02, installed in the B-sump oil scavenge system, in accordance with Paragraphs 3A through 3B of the Accomplishment Instructions of GE ASB CF34-AL S/B 79-A0014, Revision 3, dated January 31, 2003; and ASB CF34-BJ S/B 79-A0015, Revision 3, dated January 31, 2003; and the following:

(1) At intervals not to exceed 200 hours time-since-last-inspection (TSLI), if no coke is found in screens during initial or any prior inspections, or

(2) At intervals not to exceed 100 hours TSLI, if coke is found in screens during initial or any prior inspections.

Terminating Actions

(c) Install new screenless fittings or fittings that have been reworked to remove the

screens, in the B-sump oil scavenge system, in accordance with GE ASB CF34-AL S/B 79-A0016, dated June 17, 2002; or ASB CF34-BJ S/B 79-A0017, dated June 17, 2002, and the following schedule:

(1) For engines with more than 4,000 hours TSN, within 500 hours TIS after the effective date of this AD, or within 1,000 hours TSLSV, whichever occurs first.

(2) For engines with less than or equal to 4,000 hours TSN, prior to 4,500 hours TSN.

This constitutes terminating action to the inspections required in paragraph (b) of this AD.

Alternative Methods of Compliance

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Engine Certification Office (ECO). Operators must submit their request through an appropriate

FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, ECO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this airworthiness directive, if any, may be obtained from the ECO.

Special Flight Permits

(e) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be done.

Documents That Have Been Incorporated by Reference

(f) The actions must be done in accordance with the following General Electric Company Alert Service Bulletins (ASB):

Document No.	Pages	Revision	Date
ASB CF34-AL S/B 79-A0014 Total Pages: 10	All	3	January 31, 2003.
ASB CF34-BJ S/B 79-A0015 Total Pages: 9	All	3	January 31, 2003.
ASB CF34-AL S/B 79-A0016 Total Pages: 12	All	Original	June 17, 2002.
ASB CF34-BJ S/B 79-A0017 Total Pages: 11	All	Original	June 17, 2002.

This incorporation by reference of General Electric Alert Service Bulletin (ASB) CF34-AL S/B 79-A0014, Revision 3, dated January 31, 2003; ASB CF34-BJ S/B 79-A0015, Revision 3, dated January 31, 2003; GE ASB CF34-AL S/B 79-A0016 and ASB CF34-BJ S/B 79-A0017, both dated June 17, 2002, was approved by the Director of the Federal Register on April 2, 2003, (68 FR 12086; March 18, 2003) in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from GE Aircraft Engines, 1000 Western Avenue, Lynn, MA 01910; Attention: CF34 Product Support Engineering, Mail Zone: 34017; telephone (781) 594-6323; fax (781) 594-0600. Copies may be inspected at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

Effective Date

(g) This amendment becomes effective on November 18, 2003.

Issued in Burlington, Massachusetts, on October 6, 2003.

Francis A. Favara,

Acting Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 03-25864 Filed 10-10-03; 8:45 am]

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DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. 2002-CE-58-AD; Amendment 39-13335; AD 2003-21-01]

RIN 2120-AA64

Airworthiness Directives; Univair Aircraft Corporation Models Alon A-2 and A2-A; ERCO 415-C, 415-CD, 415-D, 415-E, and 415-G; Forney F-1 and F-1A; and Mooney M10 Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: The FAA supersedes Airworthiness Directive (AD) 94-18-04