

**Inspection to Determine Modification**

(b) Within 1,000 flight hours or 18 months after the effective date of this AD, whichever occurs first, inspect the right and left AOA sensor vanes to determine whether modification (MOD) 'J' has been incorporated. Instead of inspecting the sensors, a review of airplane maintenance records is acceptable if the MOD level of the sensor can be positively determined from that review. If MOD 'J' has been incorporated in both sensors, no further action is required by this paragraph.

**Movement Tests**

(c) For any AOA sensor vane that does not have MOD 'J' installed: Prior to further flight following the inspection required by paragraph (b) of this AD, do a movement test of the AOA sensor vane per the service bulletin.

(d) If the result of the movement test in paragraph (c) of this AD is less than 110 grams, repeat the movement test prior to the accumulation of 5,000 flight hours or 24 months after accomplishing the initial test, whichever occurs first. Do the test per the service bulletin.

**Corrective Action**

(e) If the result of any movement test in paragraph (c) or paragraph (d) of this AD is 110 grams or more, replace the AOA sensor vane with a reworked MOD 'J' sensor vane, per the service bulletin, at the applicable time in paragraph (e)(1), (e)(2), or (e)(3) of this AD.

(1) If the result of the movement test in paragraph (c) of this AD is between 110 and 169 grams inclusive, replace the sensor vane at the earlier of 1,000 flight hours, or 18 months after accomplishing the movement test in paragraph (c) of this AD.

(2) If the result of any repeat movement test in paragraph (d) of this AD is between 110 and 169 grams inclusive, replace the sensor vane at the earlier of 1,000 flight hours or 6 months after accomplishing the movement test in paragraph (d) of this AD.

(3) If the result of the movement test is 170 grams or more, replace the sensor vane within 5 days after the accomplishing the movement test in paragraph (c) or paragraph (d) of this AD.

**Parts Installation**

(f) As of the effective date of this AD, no person may install a sensor vane, part number 861CAB, on any airplane unless MOD "J" has been incorporated.

**Reporting and Parts Modification**

(g) Although the Rosemount service bulletin contains procedures for sending test findings to the manufacturer, and for sending removed parts to the manufacturer for modification, this AD does not require those actions.

**Actions Accomplished Per Previous Release of Service Bulletin**

(h) Actions accomplished before the effective date of this AD per Bombardier Alert Service Bulletin A8-27-94, dated October 25, 2001, are considered acceptable for compliance with the corresponding action specified in this AD.

**Alternative Methods of Compliance**

(i) In accordance with 14 CFR 39.19, the Manager, New York Aircraft Certification Office, FAA, is authorized to approve alternative methods of compliance for this AD.

**Incorporation by Reference**

(j) Unless otherwise specified in this AD, the actions shall be done in accordance with Bombardier Alert Service Bulletin A8-27-94, Revision "A", dated February 5, 2002. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Bombardier, Inc., Bombardier Regional Aircraft Division, 123 Garratt Boulevard, Downsview, Ontario M3K 1Y5, Canada. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, New York Aircraft Certification Office, 1600 Stewart Avenue, suite 410, Westbury, New York; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

**Note 2:** The subject of this AD is addressed in Canadian airworthiness directive CF-2001-46, dated December 3, 2001.

**Effective Date**

(k) This amendment becomes effective on August 3, 2004.

Issued in Renton, Washington, on June 16, 2004.

**Ali Bahrami,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 04-14319 Filed 6-28-04; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

**[Docket No. 2003-NM-236-AD; Amendment 39-13690; AD 2004-13-08]**

**RIN 2120-AA64**

**Airworthiness Directives; Short Brothers Model SD3-60 Series Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to all Short Brothers Model SD3-60 series airplanes. This AD requires inspection of the welded joints of the balance weight brackets for the left and right elevator trim tabs for

cracking; repetitive inspections, as applicable; and corrective actions including the eventual replacement of all brackets, which constitutes terminating action for the repetitive inspections. This action is necessary to prevent the loss of the balance weight for the elevator trim tab, which could result in incorrect trim during takeoff and landing, and reduced controllability of the airplane. This action is intended to address the identified unsafe condition.

**DATES:** Effective August 3, 2004.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of August 3, 2004.

**ADDRESSES:** The service information referenced in this AD may be obtained from Short Brothers, Airworthiness & Engineering Quality, P.O. Box 241, Airport Road, Belfast BT3 9DZ, Northern Ireland. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

**FOR FURTHER INFORMATION CONTACT:**

Todd Thompson, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-1175; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all Short Brothers Model SD3-60 series airplanes was published in the **Federal Register** on April 22, 2004 (69 FR 21766). That action proposed to require inspection of the welded joints of the balance weight brackets for the left and right elevator trim tabs for cracking; repetitive inspections, as applicable; and corrective actions including the eventual replacement of all brackets, which would constitute terminating action for the repetitive inspections.

**Comments**

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

## Conclusion

We have determined that air safety and the public interest require the adoption of the rule as proposed.

## Cost Impact

We estimate that 42 airplanes of U.S. registry will be affected by this AD, that it will take approximately 12 work hours per airplane to accomplish the required inspections, and that the average labor rate is \$65 per work hour. Based on these figures, the cost impact of this action on U.S. operators is estimated to be \$32,760, or \$780 per airplane.

It will take approximately 8 work hours per airplane to accomplish the replacement of the brackets. Required parts will cost approximately \$632 per airplane. Based on these figures, the cost impact of this action on U.S. operators is estimated to be \$48,384, or \$1,152 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

## Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

## List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

## Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

## PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

### § 39.13 [Amended]

■ 2. Section 39.13 is amended by adding the following new airworthiness directive:

#### 2004–13–08 Short Brothers PLC:

Amendment 39–13690. Docket 2003–NM–236–AD.

**Applicability:** All Model SD3–60 series airplanes, certificated in any category.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent the loss of the balance weight for the elevator trim tab, which could result in incorrect trim during takeoff and landing, and reduced controllability of the airplane, accomplish the following:

#### Service Bulletin Reference

(a) The term "service bulletin," as used in this AD, means the Accomplishment Instructions of Short Brothers Service Bulletin SD360–55–20, dated June 26, 2003.

#### Initial Inspection

(b) Within 2 months after the effective date of this AD: Do a dye penetrant inspection for cracking in the welded joints of the balance weight brackets for the left and right elevator trim tabs, in accordance with the service bulletin.

#### Investigative and Corrective Actions if No Cracking Is Found

(c) If no cracking is found during the inspection required by paragraph (b) of this AD, do the actions required by paragraphs (c)(1) and (c)(2) of this AD at the applicable compliance times.

(1) Repeat the inspection required by paragraph (b) of this AD at intervals not to exceed 4,800 flight hours until the bracket is replaced per paragraph (c)(2) or (d) of this AD.

(2) Prior to the accumulation of 28,800 total flight hours, or within 6 months after the effective date of this AD, whichever occurs later: Replace any bracket that has not been replaced per paragraph (d) of this AD with a new bracket or with a serviceable bracket that has been inspected in accordance with paragraph (b) of this AD. Replace in accordance with the service bulletin. Replacement of the brackets

constitutes terminating action for the repetitive inspections required by paragraph (c)(1) of this AD.

#### Corrective Actions if Any Cracking Is Found

(d) If any cracking is found during any inspection required by paragraph (b) or (c) of this AD: Before further flight, accomplish the applicable action in paragraph (d)(1) or (d)(2) of this AD in accordance with the service bulletin.

(1) For airplanes that have accumulated less than 28,800 flight hours and on which all cracking on brackets is less than 0.25 inch in length: Repair the affected bracket in accordance with Part B of the service bulletin (including the additional dye penetrant inspection of the repaired welded joint) and repeat the inspection required by paragraph (b) of this AD at intervals not to exceed 4,800 flight hours; or replace the bracket in accordance with paragraph (d)(2) of this AD. Replacement of the bracket constitutes terminating action for the repetitive inspections.

(2) For any airplane on which any cracking on a bracket is 0.25 inch in length or greater, and for any airplane that has accumulated 28,800 flight hours or more on which any cracking of any length is found on a bracket: Replace the affected bracket with a new bracket or with a serviceable bracket that has been inspected in accordance with paragraph (b) of this AD. Replacement of the bracket constitutes terminating action for the repetitive inspections required by paragraph (d)(1) of this AD.

#### Refitting

(e) Before further flight following any inspection per paragraphs (b) or (c) of this AD; or before further flight following repair or replacement of a bracket per paragraphs (c)(2) or (d) of this AD: Refit the balance weights, covers, and trim tabs, in accordance with the service bulletin. Where the service bulletin specifies to contact the manufacturer for disposition of certain conditions while refitting, obtain further disposition instructions from the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate; or the Civil Aviation Authority (CAA) (or its delegated agent).

#### Parts Installation

(f) As of the effective date of this AD, no person may install on any airplane a balance weight bracket unless the welded joint has been inspected in accordance with paragraph (b) of this AD.

#### Alternative Methods of Compliance

(g) In accordance with 14 CFR 39.19, the Manager, International Branch, ANM–116, is authorized to approve alternative methods of compliance for this AD.

#### Incorporation by Reference

(h) Unless otherwise specified in this AD the actions shall be done in accordance with Short Brothers Service Bulletin SD360–55–20, dated June 26, 2003. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Short Brothers, Airworthiness & Engineering Quality, P.O.

Box 241, Airport Road, Belfast BT3 9DZ, Northern Ireland. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

**Note 1:** The subject of this AD is addressed in British airworthiness directive 009-06-2003.

#### Effective Date

(i) This amendment becomes effective on August 3, 2004.

Issued in Renton, Washington, on June 16, 2004.

**Ali Bahrami,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 04-14321 Filed 6-28-04; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2001-NM-331-AD; Amendment 39-13692; AD 2004-13-10]

**RIN 2120-AA64**

#### **Airworthiness Directives; Bombardier Model DHC-8-102, -103, -106, -201, -202, -301, -311, and -315 Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to certain Bombardier Model DHC-8-102, -103, -106, -201, -202, -301, -311, and -315 airplanes, that requires rework/retrofit of the wardrobe shelf assembly. This action is necessary to prevent the wardrobe shelf and attached equipment separating from the attachment in the event of a hard landing, which could impede the egress of passengers in the event of an emergency evacuation. This action is intended to address the identified unsafe condition.

**DATES:** Effective August 3, 2004.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of August 3, 2004.

**ADDRESSES:** The service information referenced in this AD may be obtained from Bombardier, Inc., Bombardier Regional Aircraft Division, 123 Garratt

Boulevard, Downsview, Ontario M3K 1Y5, Canada. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, New York Aircraft Certification Office, 1600 Stewart Avenue, suite 410, Westbury, New York; or at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

#### FOR FURTHER INFORMATION CONTACT:

Leung Lee, Aerospace Engineer, Systems and Flight Test Branch, ANE-172, FAA, New York Aircraft Certification Office, 1600 Stewart Avenue, Westbury, suite 410, New York 11590; telephone (516) 228-7309; fax (516) 794-5531.

#### SUPPLEMENTARY INFORMATION:

A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Bombardier Model DHC-8-102, -103, -106, -201, -202, -301, -311, and -315 airplanes was published in the **Federal Register** on April 1, 2004 (69 FR 17113). That action proposed to require rework/retrofit of the wardrobe shelf assembly.

#### Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

#### Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

#### Cost Impact

The FAA estimates that 18 airplanes of U.S. registry will be affected by this AD, that it will take approximately 20 work hours per airplane to accomplish the required actions, and that the average labor rate is \$65 per work hour. Required parts will cost approximately \$1,387 per airplane. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$48,366, or \$2,687 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD

were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

#### Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

■ Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### **PART 39—AIRWORTHINESS DIRECTIVES**

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### **§ 39.13 [Amended]**

■ 2. Section 39.13 is amended by adding the following new airworthiness directive:

**2004-13-10 Bombardier, Inc.** (Formerly de Havilland, Inc.): Amendment 39-13692. Docket 2001-NM-331-AD.

*Applicability:* Model DHC-8-102, -103, -106, -201, -202, -301, -311, and -315 airplanes, serial numbers 452, 464, 490, 506,