

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

The Boeing Company: Docket No. FAA–2016–5041; Directorate Identifier 2015–NM–102–AD.

(a) Comments Due Date

We must receive comments by May 20, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to The Boeing Company Model 747–8 and 747–8F series airplanes, certified in any category, as identified in Boeing Alert Service Bulletin 747–57A2348, dated June 12, 2015.

(d) Subject

Air Transport Association (ATA) of America Code 57, Wings.

(e) Unsafe Condition

This AD was prompted by a report that static strength analysis has shown that the aluminum transmission aft bearing plate assemblies have inadequate structural strength for one or more of the required load cases, including cases for drive system jam, flap skew, and structural damage tolerance. Inadequate structural strength can result in damage to the transmission aft bearing plate assemblies. We are issuing this AD to prevent inadequate structural strength of transmission aft bearing plate assemblies. This condition could result in damaged transmission aft bearing plate assemblies, which could result in incorrect operation and departure of the flap from the airplane and consequent loss of controllability of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Replacement

Within 48 months after the effective date of this AD: Remove aluminum transmission aft bearing plate assemblies from the flap track and install new titanium transmission aft bearing plate assemblies to the flap track, in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin 747–57A2348, dated June 12, 2015.

(h) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Seattle Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19,

send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (i)(1) of this AD. Information may be emailed to: 9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair, modification, or alteration required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Seattle ACO, to make those findings. To be approved, the repair method, modification deviation, or alteration deviation must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) For service information that contains steps that are labeled as Required for Compliance (RC), the provisions of paragraphs (h)(4)(i) and (h)(4)(ii) of this AD apply.

(i) The steps labeled as RC, including substeps under an RC step and any figures identified in an RC step, must be done to comply with the AD. An AMOC is required for any deviations to RC steps, including substeps and identified figures.

(ii) Steps not labeled as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the RC steps, including substeps and identified figures, can still be done as specified, and the airplane can be put back in an airworthy condition.

(i) Related Information

(1) For more information about this AD, contact Bill Ashforth, Aerospace Engineer, Airframe Branch, ANM–120S, FAA, Seattle ACO, 1601 Lind Avenue SW., Renton, WA 98057–3356; phone: 425–917–6432; fax: 425–917–6590; email: bill.ashforth@faa.gov.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H–65, Seattle, WA 98124–2207; telephone 206–544–5000, extension 1; fax 206–766–5680; Internet <https://www.myboeingfleet.com>. For information on the availability of this material at the FAA, call 425–227–1221.

Issued in Renton, Washington, on March 24, 2016.

Michael Kaszycki,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016–07578 Filed 4–4–16; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2016–4123; Directorate Identifier 2016–NE–06–AD]

RIN 2120–AA64

Airworthiness Directives; International Aero Engines AG Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for certain International Aero Engines AG (IAE) V2522–A5, V2524–A5, V2525–D5, V2527–A5, V2527E–A5, V2527M–A5, V2528–D5, V2530–A5, and V2533–A5 turbofan engines. This proposed AD was prompted by the fracture of the high-pressure turbine (HPT) stage 2 hub during flight, which resulted in an in-flight shutdown (IFSD), undercowl fire, and smoke in the cabin. This proposed AD would require inspecting the HPT stage 1 hub and HPT stage 2 hub, and, if necessary, their replacement with parts that are eligible for installation. We are proposing this AD to prevent failure of the HPT stage 1 or HPT stage 2 hubs, which could result in uncontained HPT blade release, damage to the engine, and damage to the airplane.

DATES: We must receive comments on this proposed AD by June 6, 2016.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- **Federal eRulemaking Portal:** Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- **Fax:** 202–493–2251.

- **Mail:** U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC 20590.

- **Hand Delivery:** Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this NPRM, contact International Aero Engines AG, 400 Main Street, East Hartford, CT 06118; phone: 860–368–3700; fax: 860–368–4600; email: iaeinfo@iae2500.com; Internet: <https://www.iaeworld.com>. You may view this service information at the FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For

information on the availability of this material at the FAA, call 781-238-7125.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-4123 or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT:

Brian Kierstead, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781-238-7772; fax: 781-238-7199; email: brian.kierstead@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposal. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2016-4123; Directorate Identifier 2016-NE-06-AD" at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

We received a report of an engine IFSD and subsequent undercowl fire on an IAE V2527-A5 turbofan engine during a revenue flight of an Airbus A320 airplane in September 2014. The subsequent investigation of this event determined that it was caused by a manufacturing defect in the HPT stage 2 hub that resulted in fracture and failure of the HPT stage 2 hub. The event involved release of a fir tree lug and two HPT stage 2 blades. IAE also identified a similar manufacturing defect on the HPT stage 1 hub. This condition, if not corrected, could result

in uncontained HPT blade release, damage to the engine, and damage to the airplane.

Related Service Information Under 1 CFR Part 51

We reviewed IAE Non-Modification Service Bulletin (NMSB) No. V2500-ENG-72-0661, Revision No. 1, dated February 5, 2016. The NMSB describes procedures for inspecting the HPT stage 1 and stage 2 hubs. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

FAA's Determination

We are proposing this AD because we evaluated all the relevant information and determined the unsafe condition described previously is likely to exist or develop in other products of the same type design.

Proposed AD Requirements

This proposed AD would require inspecting the engine HPT stage 1 hub and HPT stage 2 hub, and, if necessary, their replacement with parts eligible for installation.

Costs of Compliance

We estimate that this proposed AD affects 668 engines with 947 hubs installed on airplanes of U.S. registry. Some of the 668 engines have two hubs installed. We estimate that it would take about 8 hours per hub to perform the piece-part inspection. The average labor rate is \$85 per hour. We estimate that 568 hubs will require replacement. We estimate the pro-rated cost to replace an HPT stage 1 hub to be \$50,271 and the pro-rated cost to replace an HPT stage 2 hub to be \$40,063. Based on these figures, we estimate the cost of this proposed AD on U.S. operators to be \$26,298,816.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for

safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

International Aero Engines AG: Docket No. FAA-2016-4123; Directorate Identifier 2016-NE-06-AD.

(a) Comments Due Date

We must receive comments by June 6, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to International Aero Engines AG (IAE) V2522-A5, V2524-A5,

V2525–D5, V2527–A5, V2527E–A5, V2527M–A5, V2528–D5, V2530–A5, and V2533–A5, engines with either of the following installed:

(1) High-pressure turbine (HPT) stage 1 hub, part number (P/N) 2A5001, with a serial number (S/N) listed in Table 1, Appendix A, of IAE Non-Modification Service Bulletin (NMSB) No. V2500–ENG–72–0661, Revision 1, dated February 5, 2016; or

(2) HPT stage 2 hub, P/N 2A4802, with an S/N listed in Table 2, Appendix A, of IAE NMSB No. V2500–ENG–72–0661, Revision 1, dated February 5, 2016.

(d) Unsafe Condition

This AD was prompted by the fracture of the HPT stage 2 hub during flight, which resulted in an in-flight shutdown, undercowl fire, and smoke in the cabin. We are issuing this AD to prevent failure of the HPT stage 1 or HPT stage 2 hubs, which could result in uncontained HPT blade release, damage to the engine, and damage to the airplane.

(e) Compliance

Comply with this AD within the compliance times specified, unless already done.

(1) Inspect the HPT stage 1 hub, P/N 2A5001, and HPT stage 2 hub, P/N 2A4802, at the next shop visit or as follows, whichever comes first:

(i) For hubs with 0 to 7,000 CSN, before accumulating 13,000 CSN;

(ii) For hubs with 7,001 to 11,000 CSN, within 6,000 cycles from the effective date of this AD or before accumulating 15,000 CSN, whichever occurs first;

(iii) For hubs with 11,001 to 15,500 CSN, within 4,000 cycles from the effective date of this AD or before accumulating 17,000 CSN, whichever occurs first;

(iv) For hubs with 15,501 CSN or greater, within 1,500 cycles from the effective date of this AD.

(2) Use Accomplishment Instructions, paragraphs 2.A., 2.C., and 2.D., of IAE NMSB No. V2500–ENG–72–0661, Revision 1, dated February 5, 2016, to inspect the HPT stage 1 hub, P/N 2A5001.

(3) Use Accomplishment Instructions, paragraphs 2.E., 2.G., and 2.H., of IAE NMSB No. V2500–ENG–72–0661, to inspect the HPT stage 2 hub, P/N 2A4802.

(4) Remove from service any HPT stage 1 hub, P/N 2A5001, or HPT stage 2 hub, P/N 2A4802, that fail the inspections required by paragraphs (e)(2) and (e)(3) of this AD, and replace with a part that is eligible for installation.

(f) Definition

For the purpose of this AD, a “shop visit” is the induction of an engine into the shop for maintenance involving the separation of pairs of major mating engine flanges, except that the separation of engine flanges solely for the purposes of transportation without subsequent engine maintenance does not constitute an engine shop visit.

(g) Alternative Methods of Compliance (AMOCs)

The Manager, Engine Certification Office, FAA, may approve AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to

make your request. You may email your request to: ANE-AD-AMOC@faa.gov.

(h) Related Information

(1) For more information about this AD, contact Brian Kierstead, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781–238–7772; fax: 781–238–7199; email: brian.kierstead@faa.gov.

(2) For service information identified in this proposed AD, contact International Aero Engines AG, 400 Main Street, East Hartford, CT 06118; phone: 860–368–3700; fax: 860–368–4600; email: iaeinfo@iae2500.com; Internet: <https://www.iaeworld.com>.

(3) You may view this service information at the FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781–238–7125.

Issued in Burlington, Massachusetts, on March 24, 2016.

Colleen M. D'Alessandro,

Manager, Engine & Propeller Directorate, Aircraft Certification Service.

[FR Doc. 2016–07579 Filed 4–4–16; 8:45 am]

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DEPARTMENT OF LABOR

Office of Workers' Compensation Programs

20 CFR Part 30

RIN 1240–AA08

Claims for Compensation Under the Energy Employees Occupational Illness Compensation Program Act

AGENCY: Office of Workers' Compensation Programs, Department of Labor.

ACTION: Notice of proposed rulemaking; reopening of comment period.

SUMMARY: The Department of Labor is reopening and extending the comment period for the notice of proposed rulemaking it published on November 18, 2015 (80 FR 72296). The Department originally allowed a 60-day comment period that was scheduled to close on January 19, 2016, but on that date extended the comment period another 30 days through February 18, 2016 (81 FR 2787). This notice indicates that the comment period is being reopened as of April 5, 2016 and extended for an additional period. The comment period for the information collection requirements in the proposed rule ended on December 18, 2015, and that period is not being reopened.

DATES: The comment period for the notice of proposed rulemaking published on November 18, 2015 (80 FR

72296) and extended at 81 FR 2787 (January 19, 2016) is reopened. The Department will accept written comments on the notice of proposed rulemaking from interested parties that are submitted from April 5, 2016 through May 9, 2016.

ADDRESSES: Parties may submit comments on the regulations in the proposed rule, identified by Regulatory Information Number (RIN) 1240–AA08, by any ONE of the following methods:

Federal e-Rulemaking Portal: The Internet address to submit comments on the regulations in the proposed rule is www.regulations.gov. Follow the Web site instructions for submitting comments. Comments will also be available for public inspection on the Web site.

Mail or Hand Delivery: Submit written comments by mail to Rachel P. Leiton, Director, Division of Energy Employees Occupational Illness Compensation, Office of Workers' Compensation Programs, U.S. Department of Labor, Room C–3321, 200 Constitution Avenue NW., Washington, DC 20210. The Department will only consider mailed comments that have been postmarked by the U.S. Postal Service or other delivery service on or before the deadline for comments.

Instructions: All comments must cite RIN 1240–AA08 that has been assigned to this rulemaking. Receipt of any comments, whether by Internet, mail or hand delivery, will not be acknowledged.

FOR FURTHER INFORMATION CONTACT:

Rachel P. Leiton, Director, Division of Energy Employees Occupational Illness Compensation, Office of Workers' Compensation Programs, U.S. Department of Labor, Room C–3321, 200 Constitution Avenue NW., Washington, DC 20210, Telephone: 202–693–0081 (this is not a toll-free number).

Individuals with hearing or speech impairments may access this telephone number via TTY by calling the toll-free Federal Information Relay Service at 1–800–877–8339.

SUPPLEMENTARY INFORMATION:

In response to requests from members of the public, the Department has decided to reopen the public comment period for the notice of proposed rulemaking it published on November 18, 2015 (80 FR 72296). The Department originally allowed a 60-day comment period that was scheduled to close on January 19, 2016, but on that date extended the comment period another 30 days through February 18, 2016 (81 FR 2787). The comment period is being reopened as of April 5, 2016 and extended through May 9, 2016. The comment