

the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2017–10–23 Airbus: Amendment 39–18897; Docket No. FAA–2016–9431; Directorate Identifier 2016–NM–104–AD.

(a) Effective Date

This AD is effective June 29, 2017.

(b) Affected ADs

None.

(c) Applicability

This AD applies to Airbus Model A321–111, –112, –131, –211, –212, –213, –231, and –232 airplanes, certificated in any category, all manufacturer serial numbers.

(d) Subject

Air Transport Association (ATA) of America Code 53, Fuselage.

(e) Reason

This AD was prompted by a determination from fatigue testing on the Model A321 airframe that cracks could develop on holes at certain fuselage frame locations. We are issuing this AD to detect and correct cracking at certain hole locations in the fuselage frame, which could result in reduced structural integrity of the airplane.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Repetitive Inspections

At the later of the times specified in paragraphs (g)(1) and (g)(2) of this AD: Do a special detailed (rototest) inspection for cracking of the affected holes at frame 35.2A on the left-hand side and right-hand side between stringer 22 and stringer 23, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320–53–1315, dated January 13, 2016 (right-hand side); and Airbus Service Bulletin A320–53–1316, dated January 13, 2016 (left-hand side). Repeat the inspection of the affected holes thereafter at intervals not to exceed 21,500 flight cycles or 43,100 flight hours, whichever occurs first.

(1) Before exceeding 25,400 total flight cycles or 50,900 total flight hours since first flight of the airplane, whichever occurs first.

(2) Within 3,300 flight cycles after the effective date of this AD.

(h) Repair

If any crack is found during any inspection required by paragraph (g) of this AD: Before further flight, repair using a method approved by the Manager, International

Branch, ANM–116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). Although the service information specified in paragraph (g) of this AD specifies to contact Airbus for repair instructions, and specifies that action as “RC” (Required for Compliance), this AD requires repair as specified in this paragraph. Repair of an airplane as required by this paragraph does not constitute terminating action for the repetitive actions required by paragraph (g) of this AD, unless specified otherwise in the instructions provided by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or EASA; or Airbus's EASA DOA.

(i) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Sanjay Ralhan, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057–3356; telephone 425–227–1405; fax 425–227–1149. Information may be emailed to: 9-ANM-116-AMOC-REQUESTS@faa.gov. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(2) *Contacting the Manufacturer:* For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or EASA; or Airbus's EASA DOA. If approved by the DOA, the approval must include the DOA-authorized signature.

(3) *Required for Compliance (RC):* Except as required by paragraph (h) of this AD: If any service information contains procedures or tests that are identified as RC, those procedures and tests must be done to comply with this AD; any procedures or tests that are not identified as RC are recommended. Those procedures and tests that are not identified as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the procedures and tests identified as RC can be done and the airplane can be put back in an airworthy condition. Any substitutions or changes to procedures or tests identified as RC require approval of an AMOC.

(j) Related Information

Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA Airworthiness Directive 2016–0106, dated

June 6, 2016, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2016–9431.

(k) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(i) Airbus Service Bulletin A320–53–1315, dated January 13, 2016.

(ii) Airbus Service Bulletin A320–53–1316, dated January 13, 2016.

(3) For service information identified in this AD, contact Airbus, Airworthiness Office—ELAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>.

(4) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425–227–1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on May 10, 2017.

Jeffrey E. Duven,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2017–10264 Filed 5–24–17; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2017–0048; Directorate Identifier 2016–CE–035–AD; Amendment 39–18876; AD 2017–10–02]

RIN 2120–AA64

Airworthiness Directives; Slingsby Aviation Ltd. Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are superseding airworthiness directive (AD) 2015–11–01 for Slingsby Aviation Ltd. Models T67M260 and T67M260–T3A airplanes. This AD results from mandatory continuing airworthiness information

(MCAI) issued by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as failure of a brake master cylinder pivot pin, which could cause the rudder pedal mechanism to detach from the brake cylinder. We are issuing this AD to require actions to address the unsafe condition on these products.

DATES: This AD is effective June 29, 2017.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of June 29, 2017.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0048; or in person at the Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

For service information identified in this AD, contact Marshall Aerospace and Defence Group, The Airport, Newmarket Road, Cambridge, CB5 8RX, UK; telephone: +44 (0) 1223 399856; fax: +44 (0) 7825365617; email: mark.bright@marshalladg.com; Internet: www.marshalladg.com. You may view this referenced service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148. It is also available on the Internet at <http://www.regulations.gov> by searching for Docket No. FAA-2017-0048.

FOR FURTHER INFORMATION CONTACT: Jim Rutherford, Aerospace Engineer, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329-4165; fax: (816) 329-4090; email: jim.rutherford@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to Slingsby Aviation Ltd. Models T67M260 and T67M260-T3A airplanes. That NPRM was published in the **Federal Register** on February 7, 2017 (82 FR 9537), and proposed to supersede AD 2015-11-01, Amendment 39-18164 (80 FR 30136, May 27, 2015) (“AD 2015-11-01”).

Since we issued AD 2015-11-01, new service information was issued to revise the inspection instructions and to add a new initial inspection period after

replacement of the brake master cylinder pivot pins.

The NPRM proposed to correct an unsafe condition for the specified products and was based on mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country. The MCAI states that:

An occurrence was reported where pivot pin Part Number (P/N) T67M-45-539, of rudder pedal assembly #4, installed on the right hand (RH) side of the aeroplane (RH seat, RH pedal) failed during taxi. This caused the rudder pedal mechanism to detach from the brake master cylinder.

This condition, if not detected and corrected, could cause the rudder linkages to rotate out of their normal orientation, possibly resulting in jammed rudder controls and consequent loss of control of the aeroplane.

To address this potential unsafe condition, Slingsby Advanced Composites Ltd, trading as Marshall Aerospace and Defence Group (hereafter called “Marshall” in this [EASA] AD) issued Service Bulletin (SB) SBM 200 to provide inspection instructions.

Consequently, EASA issued Emergency AD 2015-0065-E to require repetitive inspections of the brake cylinder pivot pins of rudder pedal assemblies #1 and #4 and, depending on findings, replacement of the affected pivot pin(s).

Since that [EASA] AD was issued, Marshall published SBM 200 Revision 2 to revise the inspection instructions and to introduce a new initial inspection period after replacement of brake master cylinder pivot pins on an aeroplane.

For the reason described above, this [EASA] AD retains the requirements of EASA AD 2015-065-E, which is superseded, but requires the use of the revised inspection instructions. This [EASA] AD also allows deferring the next due inspection after replacement of the pins.

The MCAI can be found in the AD docket on the Internet at: <https://www.regulations.gov/document?D=FAA-2017-0048-0002>.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM or on the determination of the cost to the public.

Conclusion

We reviewed the relevant data and determined that air safety and the public interest require adopting the AD as proposed except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

Related Service Information

Slingsby Aviation Ltd, trading as Marshall Aerospace and Defence Group has issued Marshall Aerospace and Defence Group Service Bulletin SBM 200, Revision 2, dated December 2015. The service bulletin describes procedures for inspection of the brake master cylinder pivot pin. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

Costs of Compliance

We estimate that this AD will affect 3 products of U.S. registry. We also estimate that it would take about 6 work-hours per product to comply with the basic requirements of this AD. The average labor rate is \$85 per work-hour. Required parts would cost about \$50 per product.

Based on these figures, we estimate the cost of the AD on U.S. operators to be \$1,680, or \$560 per product.

In addition, we estimate that any necessary follow-on actions would take about .5 work-hour and require parts costing \$100, for a cost of \$142.50 per product. We have no way of determining the number of products that may need these actions.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and

responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

(1) Is not a “significant regulatory action” under Executive Order 12866,

(2) Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),

(3) Will not affect intrastate aviation in Alaska, and

(4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2017–0048; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains the NPRM, the regulatory evaluation, any comments received, and other information. The street address for the Docket Office (telephone (800) 647–5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by removing Amendment 39–18164 (80 FR 30136; May 27, 2015) and adding the following new AD:

2017–10–02 Slingsby Aviation Ltd.:

Amendment 39–18876; Docket No. FAA–2017–0048; Directorate Identifier 2016–CE–035–AD.

(a) Effective Date

This airworthiness directive (AD) becomes effective June 29, 2017.

(b) Affected ADs

This AD supersedes AD 2015–11–01, Amendment 39–18164 (80 FR 30136, May 27, 2015) (“AD 2015–11–01”).

(c) Applicability

This AD applies to Slingsby Aviation Ltd. Models T67M260 and T67M260–T3A airplanes, all serial numbers, certificated in any category.

(d) Subject

Air Transport Association of America (ATA) Code 27: Flight Controls.

(e) Reason

This AD was prompted by mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as failure of a brake master cylinder pivot pin, which could cause the rudder pedal mechanism to detach from the brake cylinder. We are issuing this AD to detect and correct discrepancies of the brake master cylinder pivot pin, which could lead to detachment of the rudder pedal mechanism from the brake master cylinder with consequent loss of control.

(f) Actions and Compliance

Unless already done, do the following actions in paragraphs (f)(1) through (3) of this AD:

(1) Within 300 hours time-in-service (TIS) after June 29, 2017 (the effective date of this AD) or within 300 hours TIS after the last inspection required by AD 2015–11–01, whichever occurs first, and repetitively thereafter at intervals not to exceed 300 hours TIS or 12 months, whichever occurs first, inspect the brake master cylinder pivot pins part number (P/N) T67M–45–539 installed on rudder pedal assemblies number 1 and number 4. Do this action following paragraph C. INSPECTION of the Accomplishment Instructions in Marshall Aerospace and Defense Group Service Bulletin SBM 200, Revision 2, dated December 2015 (“SBM 200, Revision 2”).

(2) If any cracking or distortion of the brake master cylinder pivot pins is found or the pivot pin fails the dimensional check during any of the inspections required in paragraph (f)(1) of this AD, before further flight, replace the affected pivot pin with a serviceable part following paragraph C. INSPECTION of the Accomplishment Instructions in SBM 200, Revision 2.

(3) Replacement of the brake master cylinder pivot pins as required by paragraph (f)(2) of this AD does not terminate the repetitive inspections required by paragraph (f)(1) of this AD. If both brake master cylinder pivot pins are replaced at the same time, the first repetitive inspection after replacement of the pivot pins can be deferred until 1,000 hours TIS after replacement of the pivot pins.

(g) Credit for Actions Accomplished in Accordance With Previous Service Information

This AD provides credit for any inspections required in paragraph (f)(1) of this AD if completed before June 29, 2017 (the effective date of this AD) following the Accomplishment Instructions of Marshall Aerospace and Defense Group Service Bulletin SBM 200, Revision 1, dated April 2015.

(h) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, Standards Office, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Jim Rutherford, Aerospace Engineer, FAA, Small Airplane Directorate, 901 Locust, Room 301, Kansas City, Missouri 64106; telephone: (816) 329–4165; fax: (816) 329–4090; email: jim.rutherford@faa.gov. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO.

(2) *Airworthy Product*: For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(i) Related Information

Refer to MCAI EASA AD 2016–0214, dated October 27, 2016, for related information. The MCAI can be found in the AD docket on the Internet at: <https://www.regulations.gov/document?D=FAA-2017-0048-0002>.

(j) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Marshall Aerospace and Defense Group Service Bulletin SBM 200, Revision 2, dated December 2015.

(ii) Reserved.

(3) For service information identified in this AD, contact Marshall Aerospace and Defence Group, The Airport, Newmarket Road, Cambridge, CB5 8RX, UK; telephone: +44 (0) 1223 399856; fax: +44 (0) 7825365617; email: mark.bright@marshalladg.com; Internet: www.marshalladg.com.

(4) You may view this service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329–4148. In addition, you can access this service information on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2017–0048.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Kansas City, Missouri, on May 3, 2017.

Melvin Johnson,

*Acting Manager, Small Airplane Directorate,
Aircraft Certification Service.*

[FR Doc. 2017-10403 Filed 5-24-17; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 95

[Docket No. 31138; Amdt. No. 533]

IFR Altitudes; Miscellaneous Amendments

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts miscellaneous amendments to the required IFR (instrument flight rules) altitudes and changeover points for certain Federal airways, jet routes, or direct routes for which a minimum or maximum en route authorized IFR altitude is prescribed. This regulatory action is needed because of changes occurring in the National Airspace System. These changes are designed to provide for the safe and efficient use of the navigable airspace under instrument conditions in the affected areas.

DATES: Effective Date: 0901 UTC, June 22, 2017.

FOR FURTHER INFORMATION CONTACT: Thomas J Nichols, Flight Procedure Standards Branch (AMCAFS-420), Flight Technologies and Programs Division, Flight Standards Service, Federal Aviation Administration, Mike Monroney Aeronautical Center, 6500 South MacArthur Blvd. Oklahoma City,

OK. 73169 (Mail Address: P.O. Box 25082 Oklahoma City, OK. 73125) telephone: (405) 954-4164.

SUPPLEMENTARY INFORMATION: This amendment to part 95 of the Federal Aviation Regulations (14 CFR part 95) amends, suspends, or revokes IFR altitudes governing the operation of all aircraft in flight over a specified route or any portion of that route, as well as the changeover points (COPs) for Federal airways, jet routes, or direct routes as prescribed in part 95.

The Rule

The specified IFR altitudes, when used in conjunction with the prescribed changeover points for those routes, ensure navigation aid coverage that is adequate for safe flight operations and free of frequency interference. The reasons and circumstances that create the need for this amendment involve matters of flight safety and operational efficiency in the National Airspace System, are related to published aeronautical charts that are essential to the user, and provide for the safe and efficient use of the navigable airspace. In addition, those various reasons or circumstances require making this amendment effective before the next scheduled charting and publication date of the flight information to assure its timely availability to the user. The effective date of this amendment reflects those considerations. In view of the close and immediate relationship between these regulatory changes and safety in air commerce, I find that notice and public procedure before adopting this amendment are impracticable and contrary to the public interest and that good cause exists for making the amendment effective in less than 30 days.

Conclusion

The FAA has determined that this regulation only involves an established body of technical regulations for which frequent and routine amendments are necessary to keep them operationally current. It, therefore—(1) is not a “significant regulatory action” under Executive Order 12866; (2) is not a “significant rule” under DOT Regulatory Policies and Procedures (44 FR 11034; February 26, 1979); and (3) does not warrant preparation of a regulatory evaluation as the anticipated impact is so minimal. For the same reason, the FAA certifies that this amendment will not have a significant economic impact on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 95

Airspace, Navigation (air).

Issued in Washington, DC on May 19, 2017.

John Duncan,

Director, Flight Standards Service.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, part 95 of the Federal Aviation Regulations (14 CFR part 95) is amended as follows effective at 0901 UTC, June 22, 2017.

■ 1. The authority citation for part 95 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40103, 40106, 40113, 40114, 40120, 44502, 44514, 44719, 44721.

■ 2. Part 95 is amended to read as follows:

Revisions to IFR Altitudes & Changeover Point Amendment 533 Effective Date June 22, 2017

FROM	TO	MEA	MAA
§ 95.3000 Low Altitude RNAV Routes			
§ 95.3257 RNAV Route T257 Is Amended by Adding			
VENTURA, CA VOR/DME	SAN MARCUS, CA VORTAC	6300	17500
SAN MARCUS, CA VORTAC	MORRO BAY, CA VORTAC	7300	17500
MORRO BAY, CA VORTAC	CALIS, CA FIX	4100	17500
CALIS, CA FIX	BLANC, CA FIX	3400	17500
BLANC, CA FIX	HNNTR, CA WP	6600	17500
HNNTR, CA WP	DUBSS, CA WP	7000	17500
DUBSS, CA WP	CAATE, CA WP	6900	17500
CAATE, CA WP	CHAWZ, CA WP	3900	17500
CHAWZ, CA WP	PORTE, CA FIX	4200	17500
PORTE, CA FIX	THHEO, CA WP	4200	17500
THHEO, CA WP	JAMIN, CA WP	4300	17500
JAMIN, CA WP	POINT REYES, CA VOR/DME	4300	17500
POINT REYES, CA VOR/DME	FREES, CA FIX	3500	17500
FREES, CA FIX	NACKI, CA WP	4900	17500
NACKI, CA WP	MENDOCINO, CA VORTAC	5600	17500
MENDOCINO, CA VORTAC	MERRI, CA FIX	5600	17500
MERRI, CA FIX	FLUEN, CA FIX	5700	17500
FLUEN, CA FIX	PLYAT, CA FIX	6800	17500