

(4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**Honeywell International Inc.:** Docket No. FAA-2017-0034; Directorate Identifier 2016-NE-32-AD.

#### (a) Comments Due Date

We must receive comments by August 4, 2017.

#### (b) Affected ADs

None.

#### (c) Applicability

This AD applies to Honeywell International Inc. (Honeywell) AS907-1-1A turbofan engines with second stage low-pressure turbine (LPT2) rotor blades, part number (P/N) 3035602-1, installed.

#### (d) Subject

Joint Aircraft System Component (JASC) Code 7250, Turbine Section.

#### (e) Unsafe Condition

This AD was prompted by reports of loss of power due to failure of the LPT2 blade. We are issuing this AD to prevent failure of the LPT2 blades, failure of one or more engines, and loss of the airplane.

#### (f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(1) For LPT2 rotor blades, P/N 3035602-1 that have more than 8,000 hours since new on the effective date of this AD, perform a one-time borescope inspection for wear of the Z gap contact area at the blade tip shroud for each of the 62 LPT2 rotor blades within 200 hours time in service after the effective date of this AD.

(2) Use the Accomplishment Instructions, Paragraph 3.B.(1), of Honeywell Service Bulletin (SB) AS907-72-9067, Revision 1, dated March 20, 2017, to do the inspection.

(3) If the measured wear and/or fretting of any Z gap contact area is greater than 0.005 inch, replace the LPT2 rotor assembly with a part eligible for installation before further flight.

(4) Do the following actions within 200 hours time in service after the effective date of this AD:

(i) Using a borescope make a clear digital image of the Z gap contact area at the blade tip shroud of the 62 LPT2 rotor blades.

(ii) Identify the three Z gap contact areas with the greatest amount of wear and/or fretting.

(iii) Record the blade position on the LPT2 rotor assembly and the measured wear of the three Z gap contact areas with the greatest amount of wear and/or fretting.

(iv) Send the results to Honeywell at [engine.reliability@honeywell.com](mailto:engine.reliability@honeywell.com) within 30 days after completing these actions.

#### (g) Credit for Previous Actions

You may take credit for the actions required by paragraphs (f)(1) and (4) of this AD, if you performed these actions before the effective date of this AD using Honeywell SB AS907-72-9067, Revision 0, dated December 12, 2016.

#### (h) Paperwork Reduction Act Burden Statement

A federal agency may not conduct or sponsor, and a person is not required to respond to, nor shall a person be subject to a penalty for failure to comply with a collection of information subject to the requirements of the Paperwork Reduction Act unless that collection of information displays a current valid OMB Control Number. The OMB Control Number for this information collection is 2120-0056. Public reporting for this collection of information is estimated to be approximately 5 minutes per response, including the time for reviewing instructions, completing and reviewing the collection of information. All responses to this collection of information are mandatory. Comments concerning the accuracy of this burden and suggestions for reducing the burden should be directed to the FAA at: 800 Independence Ave. SW., Washington, DC 20591, Attn: Information Collection Clearance Officer, AES-200.

#### (i) Alternative Methods of Compliance (AMOCs)

The Manager, Los Angeles Aircraft Certification Office, FAA, may approve AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to make your request.

#### (j) Related Information

(1) For more information about this proposed AD, contact Joseph Costa, Aerospace Engineer, Los Angeles Aircraft Certification Office, FAA, Transport Airplane Directorate, 3960 Paramount Blvd., Lakewood, CA 90712-4137; phone: 562-627-5246; fax: 562-627-5210; email: [joseph.costa@faa.gov](mailto:joseph.costa@faa.gov).

(2) Honeywell SBs AS907-72-9067, Revision 0, dated December 12, 2016 and AS907-72-9067, Revision 1, dated March 20, 2017, can be obtained from Honeywell International Inc., using the contact

information in paragraph (j)(3) of this proposed AD.

(3) For service information identified in this proposed AD, contact Honeywell International Inc., 111 S 34th Street, Phoenix, AZ 85034-2802; phone: 800-601-3099; Internet: <https://myaerospace.honeywell.com/wps/portal/lut/>.

(4) You may view this service information at the FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

Issued in Burlington, Massachusetts, on June 13, 2017.

**Robert J. Ganley,**

*Acting Manager, Engine & Propeller Directorate, Aircraft Certification Service.*

[FR Doc. 2017-12561 Filed 6-19-17; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2017-0560; Directorate Identifier 2016-NM-172-AD]

RIN 2120-AA64

#### Airworthiness Directives; Airbus Airplanes

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** We propose to adopt a new airworthiness directive (AD) for all Airbus Model A310-203, -204, -221, -222, -304, -322, -324, and -325 airplanes. This proposed AD was prompted by reports of cracking in the drainage holes on the lower skin panel in the center wing box between frames (FR) 42 and FR 46. This proposed AD would require repetitive rotating probe inspections for cracking of the trellis boom drainage holes, the holes in the stringers bottom, and the holes of the inner pump, and corrective actions if necessary. We are proposing this AD to address the unsafe condition on these products.

**DATES:** We must receive comments on this proposed AD by August 4, 2017.

**ADDRESSES:** You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.

- *Fax:* 202-493-2251.

- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room

W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

- **Hand Delivery:** Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this NPRM, contact Airbus SAS, Airworthiness Office—EAW, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone: +33 5 61 93 36 96; fax: +33 5 61 93 44 51; email: [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet: <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0560; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

**FOR FURTHER INFORMATION CONTACT:** Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone: 425-227-2125; fax: 425-227-1149.

### SUPPLEMENTARY INFORMATION:

#### Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include “Docket No. FAA-2017-0560; Directorate Identifier 2016-NM-172-AD” at the beginning of your comments. We specifically invite comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD based on those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

### Discussion

Fatigue damage can occur locally, in small areas or structural design details, or globally, in widespread areas. Multiple-site damage is widespread damage that occurs in a large structural element such as a single rivet line of a lap splice joining two large skin panels. Widespread damage can also occur in multiple elements such as adjacent frames or stringers. Multiple-site damage and multiple-element damage cracks are typically too small initially to be reliably detected with normal inspection methods. Without intervention, these cracks will grow, and eventually compromise the structural integrity of the airplane. This condition is known as widespread fatigue damage (WFD). It is associated with general degradation of large areas of structure with similar structural details and stress levels. As an airplane ages, WFD will likely occur, and will certainly occur if the airplane is operated long enough without any intervention.

The FAA’s WFD final rule (75 FR 69746, November 15, 2010) became effective on January 14, 2011. The WFD rule requires certain actions to prevent structural failure due to WFD throughout the operational life of certain existing transport category airplanes and all of these airplanes that will be certificated in the future. For existing and future airplanes subject to the WFD rule, the rule requires that DAHs establish a limit of validity (LOV) of the engineering data that support the structural maintenance program. Operators affected by the WFD rule may not fly an airplane beyond its LOV, unless an extended LOV is approved.

The WFD rule (75 FR 69746, November 15, 2010) does not require identifying and developing maintenance actions if the DAHs can show that such actions are not necessary to prevent WFD before the airplane reaches the LOV. Many LOVs, however, do depend on accomplishment of future maintenance actions. As stated in the WFD rule, any maintenance actions necessary to reach the LOV will be mandated by airworthiness directives through separate rulemaking actions.

In the context of WFD, this action is necessary to enable DAHs to propose LOVs that allow operators the longest operational lives for their airplanes, and still ensure that WFD will not occur. This approach allows for an implementation strategy that provides flexibility to DAHs in determining the timing of service information development (with FAA approval), while providing operators with certainty

regarding the LOV applicable to their airplanes.

On April 22, 2011, we issued AD 2011-10-06, Amendment 39-16687 (76 FR 27227, May 11, 2011) (“AD 2011-10-06”), applicable to all Airbus Model A310-203, -204, -221, -222, -304, -322, -324, and -325 airplanes. That AD currently requires:

- Cold working of trellis boom drainage holes;
- Repetitive detailed or rotating probe inspections for cracking in the drainage holes on the lower skin panel in the center wing box between FR 42 and FR 46, and corrective actions if necessary, including repair; and
- Repetitive eddy current inspections for cracking of the upper corner angle fitting and the vertical tee fitting at left and right FR 40, and corrective actions if necessary, including repair and replacement of the internal angle fitting.

AD 2011-10-06 was prompted by European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, AD 2009-0057 to identify and correct an unsafe condition. The identified unsafe condition is cracking of trellis boom drainage holes, the holes in the stringers bottom, and the holes of the inner pump, which could result in reduced structural integrity of the wings.

Since issuance of AD 2011-10-06, EASA has issued EASA AD 2016-0196, dated September 30, 2016 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition for all Airbus Model A310-203, -204, -221, -222, -304, -322, -324, and -325 airplanes. The MCAI states:

DGAC France issued AD F-1992-106-132R7 to require certain inspections and modifications which addressed JAR/FAR [Joint Aviation Requirements/Federal Aviation Regulations] 25-571 requirements, related to damage-tolerance and fatigue evaluation of structure. Following the Extended Design Service Goal activities as part of the Structure Task Group for the Airbus A310 program, EASA published AD 2007-0053, which replaced DGAC France AD F-1992-106-132R7.

After EASA issued AD 2007-0053R1, the thresholds and the intervals of Airbus Service Bulletins (SB) A310-57-2050 and A310-57-2064 were updated, prompting EASA to issue AD 2009-0057 [which corresponds to FAA AD 2011-10-06] and [EASA] AD 2007-0053 was revised (R2) accordingly. EASA AD 2009-0057 also required the accomplishment of the actions specified in Airbus SB A310-57-2048 at Revision 01.

After EASA issued AD 2009-0057, in the frame of the Widespread Fatigue Damage campaign, new analysis has indicated the need for additional work included in Revision 03 of Airbus SB A310-57-2050.

For the reason described above, this new [EASA] AD retains the requirements of EASA AD 2009-0057, which is superseded, and requires inspection and corrective actions as specified in Airbus SB A310-57-2050 Revision 04.

Required actions include a repetitive rotating probe inspection for cracking of certain holes in the stringers bottom, inner pumps, and the trellis boom; and corrective actions, *i.e.*, repair of holes where cracks are discovered.

The compliance times vary depending on airplane configuration. The earliest initial inspection compliance time is 11,400 total flight cycles or 57,300 total flight hours, whichever occurs first. The latest initial compliance time is 38,700 total flight cycles or 77,500 total flight hours, whichever occurs first. The shortest repetitive interval is 6,200 flight cycles or 31,200 flight hours, whichever occurs first.

You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0560.

**Related Service Information Under 1 CFR Part 51**

We reviewed Airbus Service Bulletin A310-57-2050, Revision 04, dated March 13, 2015. This service information describes procedures for repetitive rotating probe inspections for cracking of the trellis boom drainage holes, the holes in the stringers bottom, and the holes of the inner pump, and corrective actions. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section.

**FAA’s Determination and Requirements of This Proposed AD**

This product has been approved by the aviation authority of another country, and is approved for operation in the United States. Pursuant to our bilateral agreement with the State of Design Authority, we have been notified of the unsafe condition described in the MCAI and service information

referenced above. We are proposing this AD because we evaluated all pertinent information and determined an unsafe condition exists and is likely to exist or develop on other products the same type design.

This proposed AD would not supersede AD 2011-10-06. Rather, we have determined that a stand-alone AD would be more appropriate to address the changes in the MCAI. This proposed AD would require repetitive rotating probe inspections for cracking of the trellis boom drainage holes, the holes in the stringers bottom, and the holes of the inner pump, and corrective actions, if necessary. Accomplishment of the proposed actions would then terminate the actions required by paragraph (h) of AD 2011-10-06.

**Costs of Compliance**

We estimate that this proposed AD affects 8 airplanes of U.S. registry.

We estimate the following costs to comply with this proposed AD:

**ESTIMATED COSTS**

| Action           | Labor cost                                    | Parts cost | Cost per product | Cost on U.S. operators |
|------------------|---|------------|------------------|------------------------|
| Inspection ..... | 84 work-hours × \$85 per hour = \$7,140 ..... | \$5,890    | \$13,030         | \$104,240              |

We have received no definitive data that would enable us to provide cost estimates for the on-condition actions specified in this proposed AD.

**Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA’s authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. “Subtitle VII: Aviation Programs,” describes in more detail the scope of the Agency’s authority.

We are issuing this rulemaking under the authority described in “Subtitle VII, Part A, Subpart III, Section 44701: General requirements.” Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

**Regulatory Findings**

We determined that this proposed AD would not have federalism implications

under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this proposed regulation:

1. Is not a “significant regulatory action” under Executive Order 12866;
2. Is not a “significant rule” under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**The Proposed Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

- 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

**Airbus:** Docket No. FAA-2017-0560; Directorate Identifier 2016-NM-172-AD.

**(a) Comments Due Date**

We must receive comments by August 4, 2017.

**(b) Affected ADs**

This AD affects AD 2011-10-06, Amendment 39-16687 (76 FR 27227, May 11, 2011) (“AD 2011-10-06”).

**(c) Applicability**

This AD applies to Airbus Model A310-203, -204, -221, -222, -304, -322, -324, and -325 airplanes, certificated in any category, all serial numbers.

**(d) Subject**

Air Transport Association (ATA) of America Code 57, Wings.

**(e) Reason**

This AD was prompted by reports of cracking in the drainage holes on the lower skin panel in the center wing box between frames (FR) 42 and FR 46. We are issuing this AD to detect and correct cracking of trellis boom drainage holes, the holes in the stringers bottom, and the holes of the inner pump, which could result in reduced structural integrity of the wings.

**(f) Compliance**

Comply with this AD within the compliance times specified, unless already done.

**(g) Rotating Probe Inspections and Corrective Actions**

Except as provided by paragraph (h)(1) of this AD, before exceeding the applicable threshold or grace period, whichever occurs later, as defined in paragraph 1.E., "Compliance," of Airbus Service Bulletin A310-57-2050, Revision 04, dated March 13, 2015, accomplish the rotating probe inspection for cracking of the trellis boom drainage holes, the holes in the stringers bottom, and the holes of the inner pump, as applicable, and do all applicable corrective actions, as specified in, and in accordance with the Accomplishment Instructions of Airbus Service Bulletin A310-57-2050, Revision 04, dated March 13, 2015, except as required by paragraph (h)(2) of this AD. Do all applicable corrective actions before further flight. Repeat the inspection thereafter at intervals not to exceed those defined in paragraph 1.E., "Compliance," of Airbus Service Bulletin A310-57-2050, Revision 04, dated March 13, 2015.

**(h) Exceptions to Service Information**

(1) Where Airbus Service Bulletin A310-57-2050, Revision 04, dated March 13, 2015, specifies a grace period "after receipt of the Service Bulletin without exceeding previous Service Bulletin revision values," this AD requires compliance within the specified grace period after the effective date of this AD.

(2) Where Airbus Service Bulletin A310-57-2050, Revision 04, dated March 13, 2015, specifies to contact Airbus for appropriate action, and specifies that action as "RC" (Required for Compliance): Before further flight, accomplish corrective actions in accordance with the procedures specified in paragraph (l)(2) of this AD.

**(i) No Terminating Action for Inspections**

Accomplishing corrective actions on an airplane as required by paragraph (g) or (h)(2) of this AD does not constitute terminating action for the repetitive actions required by paragraph (g) of this AD.

**(j) Terminating Action**

Accomplishment of the initial inspection required by paragraph (g) of this AD constitutes terminating action for the actions required by paragraph (h) of AD 2011-10-06.

**(k) Credit for Previous Actions**

This paragraph provides credit for actions required by paragraph (g) of this AD, if those actions were performed before the effective date of this AD using the service information

specified in Airbus Service Bulletin A310-57-2050, Revision 03, dated December 19, 2014.

**(l) Other FAA AD Provisions**

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the International Branch, send it to the attention of the person identified in paragraph (m)(2) of this AD. Information may be emailed to: [9-ANM-116-AMOC-REQUESTS@faa.gov](mailto:9-ANM-116-AMOC-REQUESTS@faa.gov). Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(2) *Contacting the Manufacturer*: Except as required by paragraph (h)(2) of this AD: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). If approved by the DOA, the approval must include the DOA-authorized signature.

(3) *Required for Compliance (RC)*: Except as required by paragraph (h)(2) of this AD: If any service information contains procedures or tests that are identified as RC, those procedures and tests must be done to comply with this AD; any procedures or tests that are not identified as RC are recommended. Those procedures and tests that are not identified as RC may be deviated from using accepted methods in accordance with the operator's maintenance or inspection program without obtaining approval of an AMOC, provided the procedures and tests identified as RC can be done and the airplane can be put back in an airworthy condition. Any substitutions or changes to procedures or tests identified as RC require approval of an AMOC.

**(m) Related Information**

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA AD 2016-0196, dated September 30, 2016, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2017-0560.

(2) For more information about this AD, contact Dan Rodina, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone: 425-227-2125; fax: 425-227-1149.

(3) For service information identified in this AD, contact Airbus SAS, Airworthiness Office—EAW, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France;

telephone: +33 5 61 93 36 96; fax: +33 5 61 93 44 51; email: [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet: <http://www.airbus.com>. You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425 227-1221.

Issued in Renton, Washington, on June 9, 2017.

**Michael Kaszycki,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 2017-12613 Filed 6-19-17; 8:45 am]

**BILLING CODE 4910-13-P**

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 71**

[Docket No. FAA-2017-0458; Airspace Docket No. 17-ASW-8]

**Proposed Amendment of Class E Airspace; Canadian, TX; and Wheeler, TX**

**AGENCY:** Federal Aviation Administration (FAA), DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This action proposes to modify Class E airspace extending upward from 700 feet above the surface at Hemphill County Airport, Canadian, TX, and Wheeler Municipal Airport, Wheeler, TX. The FAA is proposing this action due to the decommissioning of the Sayre co-located VHF omnidirectional range and tactical air navigation system (VORTAC) facility, which provided navigation guidance for the instrument procedures to these airports. The VORTAC is being decommissioned as part of the VHF omnidirectional range (VOR) Minimum Operational Network (MON) Program. This action would enhance the safety and management of instrument flight rules (IFR) operations at these airports. Additionally, the geographic coordinates of the airports would be adjusted to coincide with the FAA's aeronautical database.

**DATES:** Comments must be received on or before August 4, 2017.

**ADDRESSES:** Send comments on this proposal to the U.S. Department of Transportation, Docket Operations, 1200 New Jersey Avenue SE., West Building Ground Floor, Room W12-140, Washington, DC 20590; telephone (202) 366-9826, or 1-800-647-5527. You must identify FAA Docket No. FAA-2017-0458; Airspace Docket No. 17-ASW-8 at the beginning of your